

September 21, 1929

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By Captain Sir Hubert Wilkins

THE *Schneider Trophy Contest*



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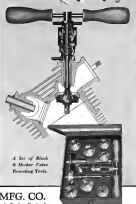


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AERO H



CONFIDENCE

THREE following excerpts were taken from the log of Capt. W. S. Lawrence covering his flight to Trinidad in an Avion powered with the Cirrus Engine. The Caribbean Circle flight has never been successfully completed and in attempt such a flight is supreme evidence of personal courage and extreme confidence in an engine.

March 3rd—Atlantic City to Hempden Roads

After two hours in the air, wind reached gale velocity and the sea boat could make no headway. Run blacking and inclement. Landed on wrong sand and in checking the Cirrus engine, found it perfect in spite of 8 hours flight in rain and worst kind of weather.

March 5th—Charleston to Jacksonville

Very soon ran into fog and flew blind for some time. Had to come within thirty feet of sea to see anything. During a poor visibility, landed on uninhabited island off the coast at 3:45 and took shelter under the machine. Traffic rule. Took off again at 2:00 P. M. The little Cirrus did not miss a revolution and on checking it at Jacksonville, found it perfect. It has been a grueling time coming from Newark in the weather I met with.

March 12th—San Juan to Guadalupe

At dawn left San Juan for Guadalupe—400 miles distant over water. Adverse weather and heavy crosswinds for 3 hours. First land I saw was Guadalupe. Looked for landing field but could see no place to be landing. Pushed the machine along that looked fairly good from air. Made good landing at midnight. Many had been down in spots.

Took off from narrow, bumpy runway made by natives for me. Only 750 feet long and more than twice as wide. Landed in capacity but the little Cirrus got off wonderfully.

These excerpts in their brevity cannot do justice to the real story of the flight but they do show the confidence encountered and the way the Cirrus engine overcame them all.

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IN severe laboratory tests this new cable has shown an ability to withstand the destructive effects of corona, oil, heat, and time that is a valuable contribution to the automotive and aircraft industries.

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COMMAND-AIRE

SPECIFICATIONS

Span (wings)	41 ft. 0 in.
Span (tail)	11 ft. 0 in.
Length	21 ft. 0 in.
Height	11 ft. 0 in.
Weight (empty)	1,100 lbs.
Weight (loaded)	1,500 lbs.
Engine (type)	Curtiss 177T P.
Engine (hp)	100
Engine (revs.)	1,800
Engine (fuel)	40 gal.
Engine (oil)	5 gal.
Engine (water)	10 gal.
Engine (air)	100 cu. ft.
Engine (fuel)	40 gal.
Engine (oil)	5 gal.
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Engine (air)	100 cu. ft.

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AVIATION

THE OLDEST AMERICAN AERONAUTICAL MAGAZINE

A MONTHLY PUBLICATION ESTABLISHED 1918

EDWARD P. WARNER, Editor

VOLUME 21 . . . September 21, 1929 . . . NO. 21



The Police Commissioner Takes the Reins

THE HEAD of New York's police department, who is never slow to seize upon a new idea, has decided to do something about aviation. To be specific, he suggests the formation of an aerial squad in the department, who will be mounted on airplanes instead of on motorcycles, and who will presumably endow some form of aerial salute to replace the familiar "pull over to the curb."

It is easy to sympathize with Commissioner Whelan's agitation, coming to it from the immediate consequence of two accidents in which innocent bystanders were killed by aircraft in the New York metropolitan area. It is not so easy to agree with him upon the desirability of police department action of the sort that he suggests. The idea is not new. It has been brooded repeatedly in the past, sometimes with serious intent and sometimes as a means of attracting public attention. New York, in fact, had an air police reserve a number of years ago, all fitted out with beautiful shiny uniforms, and other ones have toyed with the same scheme.

Rickless flying must not be tolerated. Some authority must put a stop to it—but municipal police departments cannot do the job. That simply will not work.

THE GOOD OLD DEBATE on State versus Federal control of flying was thrashed out over and over again from 1921 to 1926. The arguments in favor of Federal supremacy finally triumphed, and they remain as good today as they ever were. Every objection to State intervention applies with redoubled force against the interference of the cities in the field of aerial law-making. It is an axiom of American police administration that some of its most serious problems proceed from its being split up an state and municipal lines, and from the unwillingness of the criminal to remain in the jurisdiction in which he commits his offense. When dealing with infractions of aerial law by vehicles which pass from one side of a city

to the other in five or ten minutes all flight difficulties become prohibitive.

Regulations must be as national as the law, if they are to be generally understood and generally observed. Licensing must be solely on a national basis. If there is to be any air traffic at all, Enforcement should be national so far as practicable.

Of course it is to be taken for granted that local authorities will co-operate in enforcement work. State police especially, and certain city police officers as well, ought to have enough understanding of the regulations to recognize apparent violations and look for the Department of Commerce registration numbers on the plane. There should be no need for any special force for this work, however, and the vast wastes of money would be far better expended in increasing the scope of the Department of Commerce's activities.

THE DEPARTMENT of Commerce is always short of funds for carrying on regulations so required of it by law. The interests of the aeronautical industry will be bound to suffer from unfavorable public opinion if the ground-swinging populace does not feel that it is adequately protected against improper flying. There are two vitally necessary steps in providing such protection. First, the air regulations must be more generally understood, both by those who go into the air and by police and others who remain upon the ground. There is frequent evidence that most municipal officials are entirely ignorant of their contents, while many are unaware of their very existence. Second, the Department of Commerce in its regulatory activities must be more liberally treated by the Bureau of the Budget and by Congress, or the department must be provided with some sort of revenue of its own. These things come far ahead of Commissioner Whelan's anachronistic traffic squad in importance, in efficiency, and in probable economy.

Edward T. Jones

THE AERONAUTICAL INDUSTRY has lost a leader who could do its bidding. The death of Edward T. Jones, chief engineer of the Wright Aeronautical Corporation, following a long illness, terminates the work of one of the most important contributors to recent aircraft engine progress.

The aeronautical power-plant has had to depend upon a judicious mixture of theory and practice for its development. Mr. Jones came into the aeronautical field well fortified with theoretical background acquired during his studies and during his experience as a teacher at Cornell. He supplemented it with practical experience gained at McCook Field, and with an unflinching willingness to face the theoretical difficulties in the way of accomplishing the theoretically desirable and far better interpreted by natural law in the way of adopting the specifically attractive solution. He never feared novelty, and the creation of the high-powered air-cooled Vee engine stands largely to his credit. His work deserves to be remembered, and he has an honored place in aeronautical history. His devotion to engineering science, and his interest in young men entering the engineering field, were accorded here as he provided in his will for a substantial bequest to the technical work of the university from which he had graduated.

Considerable as was the respect everywhere felt for Mr. Jones' technical attainment, it did not exceed the personal affection of those who knew him personally. They share his family's sorrow at his too premature death.



Without Benefit of Publicity

A YEAR AGO this month a British sportsman, the Viscount de Sélincourt, and his wife started from London as a little tour by airplane. In mid-July they returned to their home, having flown a small and low-powered plane some thirty thousand miles in the interim, and having touched four continents and about a score of countries.

The Viscount and his wife were making no record. They were seeking to give no interviews. They stopped where they pleased, for as long as they pleased. They held down no special organizations along their course. They were casual tourists on pleasure bent, and they were among the passengers upon a trail which many will follow.

We earnestly upon this flight not as unique but as symbolic. What the Viscount de Sélincourt did can be done by hundreds of others. Travel by air within the United States has become a commonplace. Only a few Americans, such as Captain E. E. Aldrin and the late Lawrence Sperry and J. Dabell McKee, have made protected international tours. There should be many more

The purchaser of a two-passenger plane with an 80 hp engine who has made himself a sufficiently skilled pilot is made free of the air of the whole world. More of them should take full advantage of their opportunities. There have already been a good number of Europeans who have practiced far-flung aerial touring, like the pair who flew from London to the far East with a light plane, a suitcase, and a shakle. American sportsmen are increasing numbers should do likewise.



Meditations on the Schneider

THE SCHNEIDER RACE ends in a British triumph. Equally remarkable are the mere details to the elimination of parasitic resistance in the competing machines, the success in boosting the power of an engine already of high performance to more than double its rated power, and the smoothness with which the members of the team handled their motorized craft. Whatever regret many Americans may feel that they were unable to be represented by a team and to prove the merit of their own product and the quality of American piloting, there will be no lack of warmth in our congratulations to the British industry and to the Royal Air Force. They did the work given them, and did it with splendid success.

The completion of one Schneider race is always merely the starting point for discussion of the next. Newspaper reports after the recent race suggested the possibility that Italian competition might not continue. There has been so much innovation in official reports from Rome. Gen. Balbo has looked forward to more Schneider contests in the future, but there are many in aeronautical circles in England and in Italy as there are many in America during the years of our own competition, who feel that the drive upon national resources both in time and money is excessive, and that any good opportunity of withdrawal from governmental participation should be welcomed. A widespread conviction that the burden of annual international speed competition had become excessive was responsible in 1937 for increasing the interval between races from one to two years, and the possibility of further fundamental changes in the conditions of competition will no doubt be broached at the next meeting of the P. A. I.

One thing is certain. The Schneider race can no longer be considered as simply a sporting event. It is too costly, and the preparation for it lays too great a load upon the industry, to permit of its being regarded in that respect. The conditions under which the race is to be continued, and also the desirability of competition by any particular country, must be determined primarily from the point of view of industrial and technical benefit likely to result. Where countries have advanced, as France has done for many years and as the United States did this year and in 1937, prompt may be in-

vented, but sportsmanship is not. The decision to remove out of the competition represents only an inability on the part of government officials to see where and how the necessary expenditures would repay itself.

It is interesting to note that whether there has been much racing going on or only a 1936, and whether many nations have been actively participating or very few, the rate of upward progress of the world's speed records has been almost constant for many years. From 60 m.p.h. in 1910 to 200 in 1923 and to 360 in 1939, advance has been made, with singularly little variation, at the rate of 30 m.p.h. every two years. We agree with views recently expressed by Secretary Davies that the speed of racing machines has risen more rapidly than has the ability to put what has been learned from racing into service in practical military and commercial airplanes. We do not believe that a reduction in the number of international competitions for speed would substantially retard the progress of the art. We hope that the P. A. I. will give serious consideration to a further increase in the interval between races to three or even four years.



Census Time Is Coming

IT IS FASHIONABLE to say nowadays that business has become a science and a profession. The operations of business have attained the dignity of being taught in graduate schools. They are the object of the attention of research foundations without number.

Research has found a place in the business world because the business man has recognized his dependence on accurately known facts. Waves of depression, overproduction, unemployment crises and other such unpleasant phenomena, whether in the airplane industry or elsewhere, can be controlled and prevented only if those with power to take action have complete, reliable, and up-to-the-minute data always at their fingertips. Among manifold debts that the business world owes to President Hoover, one of the greatest is due for his efforts in providing better and prompter commercial statistics. The Aeronautics Branch of the Department of Commerce carries on nobly in the tradition that he set.

The Aeronautics Branch does what it can with the information available, but the aircraft industry still suffers a dearth of information upon its own operations and their trend. The aircraft industry, at least as much as any other, ought to inaugurate the forthcoming Census of Manufacturers and the attempts that are on foot to make it more complete than any of its predecessors.

The manufacturing census that is about to be taken will be unique. The plan for taking it and for analyzing the information gained will be reviewed and approved at every point by a committee of business men under the chairmanship of Leonard S. Horner of Connecticut, a well-known figure in the aeronautical world. Schedules

of inquiry and methods of treatment will be separately developed for each industry. The committee is convinced that the census can yield results of the utmost practical value to business men if all manufacturers will wholeheartedly coöperate. The aircraft industry ought to lead the way.



Gas and Oil Service

AS WE HAVE WATCHED the growth of air travel steadily through the Pacific States, of vast chains of magnificent highway service stations for the service of automobile vehicles, we have earnestly wished that they might serve as an example to the many aviation fields now offering only meager fueling facilities, and so facilities wherever in the way of wash rooms, telephones, or restaurants. How welcome it would be to take up to a sheltered fueling station and find a comfortable attention given to refuel and lubricate the plane while the pilot adjourned to comfortable rest rooms or an adjoining beach room.

Of course air traffic is not yet sufficiently heavy at most fields to warrant the installation of such equipment and the placing of standards purely for the servicing of aircraft. Perhaps, however, a combination plan could be worked out by which service stations designed to serve both visiting automobile parties and visiting airplane parties could be erected and operated with profit. They would attract both ground and air traffic, and should materially aid in building up the latter. Certainly the present condition at many fields where facilities consist of one empty hangar and little more is highly unsatisfactory.

One service company has erected a chain of about 30 service stations located along the Pacific Highway stretching from Canada to Mexico through the states of Washington, Oregon, and California. They are on the main highways, and are equipped with the most complete facilities for automobile servicing. Several of the stations are located on established airports and all are near good emergency landing fields. Each station is marked by a 120 ft. beacon tower with a rotating beacon light at the top and the bases of the old compass extending vertically down the tower is large red cross letters visible for many miles at night. In future they are to be developed so that all will incorporate an adjacent landing field, with complete hotel and terminal facilities for both highway and air traffic.

The development of field service facilities is one of the most vital problems in connection with the growth of aviation. Those companies which expect to reap the profits of the future must be willing to lead in the pioneering of such facilities to will greatly encourage aircraft owners to travel more widely over the air routes, both by day and by night.

SOME ASPECTS OF THE

Graf Zeppelin's

THE HEARST-ZEPPELIN globe-trotting flight established several records not likely to be broken for some time to come. The ship was not designed or built to engage in speed contests but its record flight from Lakehurst to Friedrichshafen compares so favorably with other flights that even the officers on board were astounded.

The official time in the air between Lakehurst and Friedrichshafen is 55 hours 22 min. and 45 hours 52 min. from New York to Paris. Lindbergh's time was 33 hours, and Byrd took 43 hours 21 min. from New York to the French coast.

But those of us who take part in these flights know that official time in the air and between points is not always an estimate of the performance of the machine or its navigators. Conditions vary so much that each flight must be considered on its own merit, and a comparison of time alone does not give one an accurate idea of the performance.

The circumstances on several sections of our round the world flight were favorable for making round time in an airship, and while Dr. Eckener has the best weather since that I have observed in any run, we had exceptional weather to work in. Although the usual circuitous journey was covered in such a short space of time, the zeppelin did not exceed that made during the flight from Friedrichshafen to Lakehurst, when the journey took some 82 hours.

Throughout the world flight the air speed of the airship rarely if ever exceeded 70 m.p.h. This average for the whole journey was not much more, than 50. So while the elapsed time for the first leg compares not unfavorably with airplane time we cannot lose sight of the fact that there is a great difference of speed between lighter and heavier-than-air craft. The advantage of the airship therefore is the facility to operate in bad conditions, and the consequences it affords for navigation. Airships today have considerable advantages over airplanes in the matter of navigation, but it is only reasonable to suppose that when larger airplanes with expert navigators on board do long-distance flights they will be able to take advantage of the weather as airships do today. This will speed up their time of travel in proportion to the difference of airspeed between the vehicles. Moreover, making the great facilities for navigation and the many instruments, including clock indicators, special bubble sextants, a gyro compass, and a condenser on board the Graf Zeppelin, for plotting position lines, the naviga-

tion of the world flight was not an easy matter.

It takes considerable experience to make use of the meteorological reports when they are obscure, and the information received on board the Graf was, in most cases, meager. There is really no need to be at such a disadvantage, and it is one that should be eliminated as quickly as possible. It is easy enough to get a complete weather chart from one of our friends at the meteorological office in Washington at New York, but quite a different matter to get information on board that morning enroute, and so he is in a position to know where there are good conditions of which one can take advantage.

On our flight over the Atlantic morning Aug. 8 Dr. Eckener had few reports, but in spite of this, and low visibility, he was able to take advantage of winds that increased our ground speed at times to 112 m.p.h. That point at least is 42 m.p.h. tail wind. It is easy to imagine what might have happened had we been coming the other way and unable, to avoid the wind.

Therefore I think the great lesson that has been again brought conspicuously to our notice as a result of the



The side of zeppelin after its first landing out over the Pacific.

By CAPTAIN SIR HUBERT WILKINS

(With Permission of the Royal Air Force)

FLIGHT AROUND THE WORLD

Zeppe's round-the-world flight is, thus, far more thoroughly developed the numerous logical services of the world and among for greater convenience.

An interesting observation made on this trip was that the Zeppelin could safely fly low, and for perhaps more than 50 per cent of the run, as low as an altitude less than seven hundred feet above sea level. At times we were only 300-400 ft above the ground on the sea, and its highest altitude reached was not more than 5,000 ft. This notwithstanding, the fact that we were at times between mean times 6,000-7,000 ft. high.

While I am dealing with measurements I might draw attention to the fact that the lowest temperature recorded was not over the high altitudes of Siberia, but on the way from Chicago to New York. The minimum thermometer read 42 deg. F over Siberia, 41 deg. when we were in the neighborhood of San Francisco, and a little less than 40 deg. when approaching New York. During that night was the only time we needed our bed coats blankets. As a matter of fact I was too cold to sleep, and some time in the early morning got up, put on my overcoat and got into bed again. Such conditions were to be expected at this time of the year, but most people know that the Siberian area over which we crossed is, in winter, the coldest place on earth. It is over so much colder than at the North or South Pole, where temperatures seldom go lower than 60 or 70 deg. below zero. In Siberia temperatures of 90 deg. below zero Fahrenheit has been recorded. Temperature means a great deal more to an aviator than it does to an airplane. Within reason, but temperatures are better to fly in. One can carry more gear the air is less turbulent and with really low temperatures there is usually good visibility.



The author and Dr. Eckener looking "out" in the control room of the Graf Zeppelin.

the angle of attack, only was degree to compensate by aerodynamic lift for the effect of run on the ship.

When crossing the Pacific we flew for hours in a heavy rain storm. The water just passed from the sides of the envelope. It fell up behind the floor of the control room and the place was flooded. The wind on our beam and the rain sheeted down. That was the only time on the trip when rain came on the windows or into the control room, otherwise the great bulk of the envelope sheltered the living quarters. It took hours and very foggy weather the windows clouded over, but they did not trouble the pilots at all. The officer at the wheel made use of his time by keeping a look-out at the outside and the man at the elevator control steered entirely by instruments. Apart from the few times when we were in very turbulent air we were seldom more than 1 deg. from the course but down. That would have been impossible to meet airplanes in mid-air today, but the time will come when large heavier-than-air machines will be there for an officer on watch and a helmsman who

Germany, between Friedrichshafen and Berlin we came to unexpected meteorological conditions, and I was surprised to hear Dr. Ekener say "I expect the worst about here to be in our favor." It is not as the mystery solves." But in general throughout the journey Dr. Ekener's estimate of the weather was correct. From information received he knew that it would be difficult to follow the urgent request of the Soviet Republic and cross over Moscow. His own great desire—as it was—was to fly at high altitudes over the Arctic sea. But notwithstanding his desire to go north and the Russians with that he go south. Dr. Ekener chose a middle route. When the wind was strongest in his favor, then you will find Dr. Ekener.

After leaving Friedrichshafen we flew south to Berlin, going to avoid the heavy passes for the first few hours then changing over to the ground. The change-over is effected without even stopping the engine. All that is necessary is to turn off the gasoline supply and open a check valve. There is actually any danger in the sound of the engine when the fuel is changed. After Berlin we passed over Saxony, Saxony, and over the Baltic, stopped over the corner of Latvia and Estonia and on to Riga. It was late at night when we crossed the border, and the first Russian town we saw was Wladimir. It is a city of churches where I counted 40-odd spires. The Soviet representative informed us that these churches are now used as schools. Our route took us over the low part of the Ural mountains. Their height at that point is not over 2,000 ft. They were not in themselves a very great obstacle, but the many forest fires burning in that region almost blinded us with smoke, and made it difficult to keep the course. Generally the ground could be seen, but flying as we were going from the engine was poor. Beyond the Urals we met with swamp lands and although I have seen the Barren Grounds of the Arctic, I think there is nothing more desolate in appearance than a general view of those Siberian swamps. They are probably frozen solid for six months of the year, and it is that condition would afford a good surface for traveling on foot, but it would be almost impossible to walk through them in summer time. Notwithstanding the broad desolate appearance, a close observation showed many signs of animal life and more birds in that region than at any other part of the journey. We flew low over the ground

most of the night. It was the weirdest sight I have ever seen. It was a job to tear oneself away from the window and watch a few hours sleep, but that was advisable because, fascinating as the night view might be, I could really see none of interest in daylight. As dawn I went again to the control room and found Mr. Lech asleep the ship. He was making a very good job of it. He had been up all night and told me that in about a 50-mile stretch they had seen but one small light, the only evidence of habitation.

The wind was in our favor. It carried us across the mountains and the few rolling hills that were covered with low, stunted, coniferous timber with such speed that we reached Yakutsk, 4,000 miles from our starting point in 17 hours. The estimated air distance covered was 4,800 miles, and that with only four engine reversals. Our air speed was probably never more than 60 m.p.h. and often less than 50. We averaged a 20 m.p.h. tail wind.

ONE OF THE GREAT THREATS of the journey was when we ran beneath a great ice cloud in Central Siberia. There were heavy rain squalls all about us and most of them could be dodged, but at least one had to be flown through. As we passed beneath the cloud we were sucked up as if by a living force and enveloped in the dark gray base of the cloud. On the other side we were spat out with a downward rush and lost nearly 500 ft. in a few seconds. Another fall was the approach to the Stanovoi Mountains—the great range fringing the Okhotsk Sea. There were marked on the maps as 3,400 ft. high, but even flying at 5,000 ft. we were beneath the level of the peaks. Under the guidance of Dr. Ekener, the crew steered the Zepppelin through this mountain stage as a pursuit pilot might steer his machine. We followed the valleys and the crevices, never lost beneath the mountain tops and only a few feet above the ground. Many thought we were in grave danger and wondered why we did not fly higher and over the mountain tops. But as the ship was so light by that time, and as it was always prone to spin instantaneously, Dr. Ekener, with his experience accumulated in 3,000 flights, knew he was safe. It was an amazing experience to battle down from 5,000 ft. over Asia to the edge of the Pacific Ocean, and as we did, we met with cloud and less favorable conditions. We had been notified of a cyclone in the neighborhood and most every one would have decided to alter their



A group of Zepppelin crew members in a meeting about the flight.

of it, but not our experienced German friends. If there is a mid-west anywhere they will be into it, even though it may be the dangerous end of a storm. The Graf's plans show just how far they could go with safety, and they took advantage of the storm. It is, however, a game that most had better not play, at least not until we are better provided with information as to the extent of the area involved and more certain of the behavior of our instruments. We had advantage of the storm for only a few hours. It carried us down toward Japan and into a still bled wind. The wind was not strong and in a few hours we had completed the first non-stop flight from beyond the Baltic to the Pacific. The total distance from Friedrichshafen to Kaniyama by the route we took is 6,980 miles, it was covered in 301 hours 44 min., and we still had enough fuel left to go half as far again. That was because we were able to fly on four engines most of the way—using the fifth only when backing head winds, and because of the tail winds which had helped us so materially.

The Japanese Navy gave us a free exhibition of skill and training. I have never seen men jump quicker to fulfill a command than did those Japanese boys.

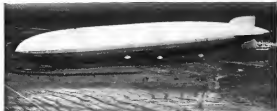
A crowd of the famous crowd of spectators showed up to see us at Tokyo but no place in an aviation hospital, except perhaps to warn those who do pioneering things that they must consider that the automobiles are likely to stop for with schedules and that it is almost impossible without roads to avoid them. However, with such a big crowd as we had on the Zepppelin there was no difficulty in maintaining the schedule, and promptly at the hour arranged we were moving out of the hangar. At that time we experienced the first slight damage sustained on the trip. It was caused by a jet in the rear engine carrying the tail fin that caused the ship, and a side, perhaps, by the fact that the guy ropes were made tight instead of being ready for quick slackening. When the order came to lift the air of the ship over an obstruction the air at the lifting horn could not budge the ship. The rear engine motor struck the ground, and

several bracing wires and some struts were broken. It might easily have been a serious accident, and some of the main girder angles have been strained. But again Zepppelin engineering proved to be adequate. The guy wires broke before doing any great damage to the main structure, and repairs were quickly effected.

Had we started that night we might have had bad weather, but it is unlikely that conditions would have been much better than they were when we actually started. Dr. Ekener has the reputation of meeting out the weather, but the weather seems to have the habit of following Ekener. Two days after he had planned to start we had to face only one small storm, and would then have good going the rest of the trip. It is good that most pilots would have avoided. The Zepppelin's officers passed the storm and decided to take it. They knew they would have to go through it from "before" that is, meet the dangerous threat of inclement as we approached the cloud, and the likelihood was small. Even as we dipped slightly enough to clear some of the passengers, and they fled to the windows. The black cloud overhead shadowed and darkened everything. For a few minutes we remained along as if crossing a series of mountains, and then the cloud broke and the ship seemed into an air stream traveling in an entirely different direction.

WITH THE SPECIAL SHIP INSURANCE, insured by the crew of the Zepppelin, they safely picked up for the new dark riding and took up the course. The dark riding generally relied upon during our flight (although the ship carried two of the usual circular head-light type) was in reality a reflex camera. The lens pointing down threw a reflection on the ground glass of the instrument, and it was in this manner so to speak that the ship's objects passed along the ruled lens upon it. The glass holder was twisted in degrees and the angle read directly from the instrument. This form of instrument was found to be exceptionally serviceable for camera operation, and is the best of its type that I have seen.

For hours during the Pacific flight we flew through thick fog where the visibility was not more than a few yards. However, through most of the daylight hours we could see the light of the blue sky above and knew that by going a few thousand feet up we could have been in clear sunshine. Many times the ship did rise above the clouds for a few observations and so a few of our pictures was obtained. At other times we would descend to within three or four hundred feet above the sea and get an idea of the angle of drift and general speed. The unfortunate thing was that the wind was of the same strength above, below, and in the fog. It was frequently



The Graf Zepppelin in the harbor of the floating city of Yokohama, N. J.

found that the ground speed had been the same throughout hours of fog flying as it was when we were beneath it taking an observation. This does not apply at all times and could not be depended upon in another flight.

WHILE FLYING in such conditions as we flew might have been difficult or almost impossible for an airplane, it would have been possible, on the day we crossed, for an airplane to have flown either beneath the clouds, depending on the compass for direction, or well above the clouds—the tops of which were not above 4,000 ft.—checking the course by sun observation. A trip by aircraft across that part of the Pacific is not by any means as easy undertaking, but in spite of so much blind flying we found that our estimated position, just before our sighting



countrywide contacting the manufacturer, the local service stations, and if necessary the engine owners. They are responsible direct to the Glenside office and segment that office with their movements from day to day. In Mr. Tucker's office there hangs a map of the United States and on its surface one can note a collection of colored pins stuck into the map at various places. One color represents the location of the service and parts depot, another color represents the location of local service stations. And still a third color shows the representative location of the field service representative.

Thus, if the factory receives an emergency call from a plane manufacturer, or a plane owner for that matter, it takes but a glance at this map to determine the field service representative nearest to the source of the emergency call. A service is immediately dispatched, if the case cannot be reached by phone, and in less than two hours the call is being "serviced." Incidentally, the field service representatives make daily reports to the Glenside office, so that the latter will be constantly acquainted with the jobs on hand in the field, and where new are available for emergency work the following day.

The field service representatives like the service and parts depot managers, are experts on Kinner engines, and therefore well qualified to apply all information desired by a manufacturer, or supervisor the satisfaction of the

sold to the plane manufacturers for redistribution to their various place distributors and plane dealers. A discount of sufficient size to enable the manufacturer to profit from the handling of the parts is allowed. Thus it will be seen that every possible means of serving Kinner owners has been, or is being, arranged.

THE MAIN SEALING MARKET for the Kinner company consisting of airplane manufacturers, the factory itself does not attempt any retail, direct-to-consumer selling. Nor does it permit its service depots, which may have an engine on hand, to try retail selling. Of course in isolated cases when a plane owner demands that the factory sell him direct, the request may be granted. However the usual procedure is for the factory to refer the customer to the manufacturer of the plane he owns and purchase his new engine through him. That is itself is a commendable policy for it is a good-will building idea with regard to the airplane manufacturer.

In fact, the officials of the Kinner company vouchsafe their successful progress from the quality of their product, to the fact that they are doing everything possible to render complete service to their customers, the airplane manufacturers, and their customers' customers, the airplane owners. Complete co-operation between the Kinner company and the airplane manufacturers using Kinner products as drivers for at all times. When a plane is sold the engine manufacturer is notified, and then the Glenside factory writes direct to the owner and acquaints him with the location of the nearest Kinner service station where he may obtain expert Kinner service. The parts depot in that territory is also notified, and it turns the local service station. The latter contacts the plane owner and the result is a satisfied owner, and satisfied owners result in increased sales for both the airplane and the engine manufacturer.

Last but not least of the Kinner company's efforts to render complete and desired service to owners of Kinner engines is the practice followed in the matter of labor charges for repair work at all Kinner service stations. In the early days of the automobile, as fast as to very long ago as that, the car owner who entered an automotive service station, or repair garage, did so at the risk of losing the entire contents of his wallet. In short, he was virtually at the mercy of the repair man as regards the cost of any repairs. After years of "suffering" on the part of the car owners the automotive industry took steps to remedy the situation, with the result that today there are more or less standard rates for various repair jobs on an automobile. However, it is still enough of a problem to be a thorn in the side of the automotive industry.

Although aero engine repairing has not as yet reached a stage where standardized charges for minor jobs can be determined, or more than just a temporary loan, the Kinner company has taken advantage of the lesson learned in the automotive industry and has set a definite hourly labor charge for repair work. Material costs naturally vary, but so matter what Kinner service station does the repair job, whether it be in New York City, or Los Angeles, the customer is charged the same flat rate per hour for labor.

When an organization stands ready at all times to render complete service to its product, and in addition takes steps to prevent the owner from being "gypped" when service is necessary, that organization is building up a clientele of satisfied customers. And satisfied customers are stepping stones to bigger and better sales.

Gliding Activities AT THE NATIONAL AIR RACES



W. E. Pringle, standing to the land during the National Air Races at Cleveland after a low from Tipton, Ohio.

AMONG the several new trends in American aeronautics displayed before the 400,000 or more persons crowding the recent National Air Races at Cleveland, few were more thoughtfully novel to a majority of the spectators than was the long-dashed sport of motorless flight.

Remorse of the fact that the Cleveland meet introduced gliding on its regular program as the first time the sport and science had been featured on the regular list of any aeronautical event of any size anywhere in this country, it is safe to assume that most spectators of the madmen got their initial sight of a glider in flight during the 10 days of competitive activity.

Despite being pushed about the program somewhat as a member might corner "sissy" off the parlor floor when "japs" is about to perform a difficult gymnastic feat, glider devotees managed to come in for their share of publicity and clapped up two major accomplishments—even though one of these was somewhat mis-carried by an unfortunate mishap of other business.

We speak of the second annual National Glider Conference, held by the National Glider Association at Elrod, Wisconsin on the morning of Tuesday, Aug. 27, and of the National Glider Contest, staged at the airport immediately following that conference and during the forenoon of the following day. In the Conference follows

of motorless flight succeeded in "pulling a fast one" on the power-driven boys by holding their meeting so early in the morning that most of the motor pilots were snuggly snoring. Otherwise, the National Glider Contesters might not have enjoyed the success it attained. As it was a good-sized

crowd attended and learned the dangers of many lessons, both in the power-driven and the motorless divisions of the craft. It took real talent to turn out as right in the morning, especially after late festivities the previous evening.

THE CONFERENCE was called to order by Clifford W. Henderson, managing director of the Races. He introduced Edward S. Evans, president of the National Glider Association, who has done much toward popularizing the sport during the past year. Following an address of welcome by Mayor John B. Mendenhall of Cleveland, Mr. Evans, who presided, outlined both the accomplished and the intended program of the N.G.A. Probably the most interesting portion of Mr. Evans' address consisted of an appeal to the aircraft industry to give assistance to the association in accomplishing its avowed purpose of making one million thrillless-temper glider pilots.

"The National Glider Association," Mr. Evans stated,



The Kinner B-3 1100 hp radial engine.

engine in place as well as any type of repair work necessary. In short they are what the tide implies — representatives of the Glenside office had an additional hand between the plane manufacturer and the engine manufacturer.

For emergency calls that come into the service and parts depots, from either the local service stations or the plane owners, the depot maintains a plane ready for instant use. A licensed pilot is a part of each depot organization. Therefore, if the report comes in that a plane is down some place and in need of engine repair, the replacement of parts for instance, the depot plane can take off immediately with a mechanic and parts and then "save the day" for the plane owner. The plane is also used by the service and parts depot manager or assistant manager to make trips about the territory and contact the various service stations.

As a further assurance that Kinner owners will receive the full benefit of Kinner service, stocks of parts are also

"has largely completed its experimental work, except in the advanced technical field, and the association now knows what to do and how to do it." As readily as sufficient funds are available, we will carry on our program, which aims for the organization of a glider club in every city, town, and hamlet in the United States, and the training of at least one million glider pilots to their third-class license. In my judgment the Association should be provided with \$50,000 annually by the aircraft industry to help defray expenses at carrying on this work."

The Evans appeal directed attention to the potential airplane market that 1,000,000 glider pilots would stimulate, as well as to the aeronautical and aerodynamic skill contrasted to the manufacturing and operating ends of the industry by such a number of youths versed in the handling of manœuvring aircraft.

After this followed an address of welcome by Floyd J. Logan, chairman of the race committee, and an introduction of L. F. Ross, director gliding activities at the Cleveland meet. Miss Jessie Eckhart, noted woman flier and glider enthusiast, then spoke briefly, stating that she found gliding an interesting and a fascinating sport, and she would be glad to do anything within her power to foster interest in gliding in this country.

FOLLOWING short addresses of a similar trend by Charles L. Lawrence, of the Curtiss-Wright group, Capt. Frank M. Hawks, holder of the trans-continental speed record, Prof. Edward P. Warner, editor of *American*, and Dr. Wolfgang Klemperer, of the Goodyear-Zeppelin Corporation, Prof. R. E. Franklin, of the University of Michigan, was asked to give an account of his experiences during the past summer when he barometered through the East with a glider, and also to present his version of auto-towing as a method of launching gliders.

The auto-towing method was extolled by Professor Franklin as being both highly interesting and safe for well-trained pilots and crews, but he said that he could not recommend it for the inexperienced, since auto towing presented elements of danger not found in properly conducted shock cord work. On the other hand, Professor Franklin stated, when preceded by experienced gliding enthusiasts, the auto-towing method really was safer than the use of shock cord. Auto towing, he said, is merely a means of putting a motor on the glider, since the glider must be made structurally stronger. Once the glider is in the air, the motor is practically useless, the automobile should be kept in a quiet rest mode ten or ten miles in excess of the glider's take-off speed, since it is possible for a glider to take a considerable portion of

the car's weight off the rear springs, on a summer that may cause damaging consequences to the glider and its pilot. In his experiences, Professor Franklin said, he found by using a tow-rope about 200 ft. long, glider being towed at a rope 700 ft. or even 800 ft. in length would serve the purpose better.

Opportunity was given those present to ask informal questions of Dr. Klemperer, who is chairman of the N.G.A. technical committee, and every invited themselves of the privilege. One of the problems brought forth in the general discussion related to the possibilities of inducing the number of men necessary to carry on gliding work. It was complained that it is sometimes difficult to find enough men willing to "go gliding," since the work entailed is so considerable in comparison with the actual gliding experience received by individual members of the party. A number of crew reducing schemes were advanced, one of the most interesting being an automatic electrical launching device.

In addition to those already mentioned the Conference was attended by a distinguished group of Air Race pilots and many well-known figures in the industry, including Mrs. Blanche Noyes, Prof. F. W. Pavlovskis of the University of Michigan, Prof. Peter Abrams of the University of Detroit, Luther R. Bell, of the Aeronautical Character of Commerce, Lt. Col. S. J. Barruch, U.S.N., the first person to qualify for a first-class glider license in America, Ray Cooper, of the Detroit Board of Commerce, and Oscar Gross, designer of the Darmstadt tower.

As has been mentioned, the contestants in the glider contests at the airport found themselves so pressed for time and space, that no fair idea of the respective abilities of all of the participants could be gained. Therefore, the N.G.A. cannot accurately, after reviewing the events, decide that, due to the conditions under which the contests were held some of the prizes would be distributed by the judges on a basis of favoritism. There are two events which had to be cancelled for lack of time were thus given an opportunity to share in the awards. On that basis trophies were awarded in the following manner:

First first prize: To the Jackson (Mich.) Glider Club, for distance, shock cord launching method, 592 ft. Elmer Winterfield, pilot.

First second prize: To the Pioneer Gliding and Soaring Society, of Orion, Mich., for distance, shock cord launching, 499 ft. Oscar Kuhn pilot.

Second first prize: To the Cleveland Glider Club, for best time, shock cord method 10-43 sec. William Strauss, pilot.

Second second prize: To the University of Detroit



Capt. Frank Hawks in the air at the 1929 Vought 5C Races at Cleveland, Ohio.

Glider Club, for best, shock cord method, 15 sec. Kenneth H. Carr, pilot.

Third first prize: To the Jackson Glider Club, for the best time, auto-towing with return to starting point, one minute, 34 sec. and landed 5 ft. from the line.

Fourth first prize: To the Cincinnati Glider Club, for having averaged second best in both distance, shock cord method, and time, auto-towing method, under adverse circumstances, and for sportsmanship during entire meet. William Fowler, pilot.

Fifth first prize: To the Pioneer Gliding and Soaring Society for their considerable contribution to gliding activities throughout the meet, for good performance, and for splendid sportsmanship.

Sixth first prize: (For auto-towed plane only.) To Capt. Frank M. Hawks, for maximum one minute, 55.5 sec. in the air and landing 14.25 yd. from the line.

Seventh second prize: To Maj. Read G. Landis, for maximum 37 sec. in the air and landing 66.5 yd. from the line. (It should be mentioned that Captain Hawks flew the secondary, high performance glider, which had been towed to Cleveland behind an airplane and landed on the airport by his pilot, Wallace Franklin, brother of the glider designer, while Major Landis flew a primary training glider loaned from the Jackson Glider Club. However, Hawks' trial and Landis' trial the secondary glider had been worked in a faulty landing by another named pilot.)

Eighth first prize: To the University of Detroit Glider Club, for good performance in both methods of launching and for sportsmanship and for having finished a close second in elapsed time on auto-towing without previous experience.

Ninth second prize: To a group of high school boys of Des Moines, Ia., who brought a home-built glider to the Races and flew it gamely throughout the shock cord launching contests. (In addition to the trophy, the Iowa youths were presented with a new PT training glider by W. J. Scripps, president of Glidden, Inc., of Orion, Mich., manufacturer.)

Tenth first prize: To Wallace Franklin, of Ann Arbor, Mich. for the most impressive performance during the meet. (Franklin, towed to Cleveland from

Ypsilanti, Mich., behind a Waco biplane, cut loose 2,500 ft. over the airport, and glided to a beautiful landing before the stands.)

Eleventh first prize: To Dr. Wolfgang Klemperer, of the Akron Glider Club, for attempting to bring a Cooder aircraft, towed by a Goodyear Blimp to Cleveland from Akron. (Dr. Klemperer, flying at about 200 ft., had covered nearly half the distance when the tow rope parted. He continued to soar, in the direction of the Cleveland airport, however, and landed 5 miles closer to his goal, near Wallings Corner, Ohio. In his report to the association, Dr. Klemperer stated he feared a field sufficiently large but that he had difficulty in bringing his ship to earth, but eventually, rising from the ground, he was able to fly nearly the entire length of the field at an altitude of only a few feet. Upon arriving the end of the field, the pilot said, he was compelled to ease the soarer into the ground, slightly ending the shell.)

Eleventh second prize: To Capt. Frank Hawks for the longest time in the air in a glider. (Captain Hawks was towed to Cleveland by an airplane from Pettib, Mich., and cut loose from the plane so high over the airport it required more than 20 min. for him to glide to a landing.)

THUS the first national glider contests were held and the various trophies duly awarded. The ability displayed by most of the youths participating in these events held reflected how American youth is living to the idea of flight.

The Cleveland airport or any other airport is of course very unsatisfactory as a ground for glider demonstrations. Soaring flight depends upon rising currents, which are only to be counted on in the neighborhood of hills, and even straight gliding with shock-cord launching must start from an elevated point to secure any satisfactory distances. The crowds at Cleveland were interested and inspired by a demonstration that had to be given under very poor conditions for any kind of gliding except the auto-towed. Donald Waller, manager of the National Glider Association, has recommended to the organization that hereafter no more glider contests should be held on airports during any seasons-plans meet.



W. H. Franklin "auto tow on the ground" at the end of a 100 mile tow

THE OPERATION

Thompson Aeronautical

By JOHN T. NEVILL

SINCE ORGANIZATION of the company in 1927 Thompson Aeronautical Corporation has become the fourth largest air mail operator in the United States in point of miles flown daily. Although the company probably has more mail tonnage than any other operator in the nation, it also serves the second, fourth and fifth largest cities in the country, *viz.* Chicago, Detroit and Cleveland. The company's line, known officially as C. A. M. 27, connects with 16 cities in the four states, 12 of them being in Michigan. This gives Michigan the fourth-whole distinction of having more cities receiving air mail service than any other state in the Union. Another distinctive feature of C. A. M. 27 is its geographic complexity. Unlike practically all other contract air mail routes, No. 27 stretches from one large city in one state to another large city in a second state, via a third large city and 12 smaller cities in a third and a fourth state. A glimpse of the accompanying map showing the Thompson airlines spreading out, irregularly, over the lower peninsula of Michigan, with two fingers extending from Michigan to Cleveland and two from Michigan to Chicago, will give the reader some idea as to this complexity.

Still another distinguishing factor in the operation of C. A. M. 27 is the emphasis on passenger, mail and express service operated six days a week between Detroit and Cleveland. This division known as the reserve division of the company's lines was inaugurated on May 14 of this year and is the only amphibious passenger, mail and express service between two inland cities within the United States.

C. A. M. 27 can be said to be an outgrowth of C. A. M. 6 and C. A. M. 2, formerly operated between Detroit and Cleveland and Detroit and Chicago, respectively, by the Ford Motor Company beginning Feb. 15, 1926, and also of the night spur line operated by National Air Transport between Detroit and the Transcontinental trunk line at Toledo. The present operating company was organized in the summer of 1927 in Cleveland by Edwin G. Thompson, vice-president of the \$6,000,000 Thompson Products Corporation, and son of C. E. Thompson, first president of the Glenn L. Martin Company, with R. C. (Tex.) Marshall, former government mail pilot, as vice-president and general manager. Early in 1928 Thompson Aeronautical Corporation was awarded a contract to carry the mail between Chicago and certain Michigan cities, including Kalamazoo, Muskegon, Grand Rapids, Bay City and Detroit, the

routes to these cities having been sought after Mr. Marshall had made a careful survey of the lower peninsula of Michigan, with particular reference to the various industrial centers likely to furnish considerable mail tonnage to the projected lines.

The lines were assigned in three separate divisions, all of them converging at Kalamazoo. One of these divisions began at Chicago and extended to Muskegon, Mich., by way of La Porte and South Bend, Ind., Kalamazoo and Grand Rapids, Mich. Later, La Porte was eliminated from the route. A second division began at Kalamazoo and



Edwin G. Thompson,
president Thompson
Aeronautical Corp. Left:
John A. Baker, oper-
ating manager



extended to Bay City, with stops at Lansing and Saginaw. The third division also began at Kalamazoo, projecting to Detroit, via Harris Creek, Jackson and Ann Arbor.

Operation of these three lines was begun on July 17, 1928, with only five planes comprising the flying equipment. From that nucleus the Thompson mail lines have grown into what can be called six divisions operating today. Chronologically, the milestones were as follows: On Nov. 27, 1928, First was made a stop on the Kalamazoo-Bay City division, and the Kalamazoo-Detroit division was extended to include Pontiac. The following April 1—April this year—the company flew its first mail

OF

Corp. Lines

over a new division Bay City—Cleveland. This division, an overnight, lockaway, service between Bay City and the transcontinental air mail line at Cleveland, is the longest of the Thompson lines, being 236 miles in length, and having stops at Saginaw, Flint, Pontiac, Detroit and Toledo. It replaced the spur line operated by National Air Transport between Ford Airport, Detroit, and the N. A. T. trunk-line at Toledo, which ceased operation on the day the Cleveland-Thompson line began. This marked the first time the Thompson mail lines touched Cleveland, in which city was, and still is, their home office. In addition to the mail over the 236 miles of its length, this division also carries express between Detroit and Cleveland.

The next extension of service came last May 18, on which date the company inaugurated the aerial passenger,

mail and express line between Detroit and Cleveland, using three Lockheed Argonauts, and maintaining a schedule of four trips daily, each way, except Sunday. Originally, the mail carrying over this division was confined to the noon flights between Detroit and Cleveland and between Cleveland and Detroit, but on July 8 the 10 a. m. Cleveland bound plane was placed into mail service. Addition of the 10 a. m. Detroit to Cleveland plane was to incorporate a stop at Toledo, resulting in greatly increased service since it made direct connection with the out-bound N. A. T. with Colorado to Buffalo, with Chicago to Buffalo, with St. Paul and Pittsburgh, and with other contractors departing from Cleveland around noon-time.

Early last Spring it became apparent to Thompson officials that



A map showing of the routes that will then be operated by Thompson Aeronautical Corporation. Note that theneapolis "connector" of the routing

they were overlooking an opportunity for still further service to air mail users because of an unfortunate lack of connections between mail planes arriving in Chicago from points west and south and crank overnight trains between Chicago and Detroit. It seemed that a majority of the cross-air mail lines "feeding" the two overnight trains, particularly the Boeing line from the west coast, did not arrive in Chicago in time for their respective cargos to be "switched" and placed aboard the first of these trains, which tails out about 9 p. m. By placing the mail aboard the second train, leaving Chicago at 12 midnight, a fortunate delay was caused in Detroit, with a delivery of mail to 24 hours if forced in delivery in other cities of Michigan.

The Thompson company cherished this delay with one comparatively inexpensive addition to its service. They reasoned that the first clock train would require 2 hours 30 min. to reach Kalamazoo, arriving there at 11:30 p. m. One of their planes could fly it easily in 1 hour 15 min., with elimination of the stop in South Bend. The plane, therefore, could resume at the Chicago Airport long after the early train had departed from the city, pick up the incoming mail from the west, south and northwest, and continue the fast train in time to place the mail aboard it in Kalamazoo.

With lighting facilities and an emergency field system between Chicago and Detroit completed as far as Kalamazoo this plan was made easier, so, on May 27, this year, the newest of the Thomp-

Thompson Aeronautical Corp.									
SOUND TRIP—1928 TRIP TIME, CLEVELAND									
FOUR-DAY SCHEDULE									
DETROIT TO CLEVELAND									
Day	Time	Plane	Time	Plane	Time	Plane	Time	Plane	Time
1	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
2	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
3	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
4	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
CLEVELAND TO DETROIT									
1	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
2	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
3	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00
4	10:00	Argonaut	11:30	Argonaut	1:00	Argonaut	2:30	Argonaut	4:00

A reproduction of the time table of Thompson Aeronautical Corporation

son divisions was placed in operation. The Chicago-Kalamazoo overnight line Thompson officials say, will be extended to Detroit as soon as the Department of Commerce has completed installation of beacons along that route.

Thus today, we have the company operating C.A.M. 37 on six divisions, each division ranging from 100 to 230 miles in length. For convenience sake we will call the Chicago-Muskegon run No. 1, the Kalamazoo-Bay City run No. 2, the Kalamazoo-Pontiac run No. 3, the Bay City-Cleveland night run No. 4, the Chicago-Kalamazoo night run No. 5, and the Detroit-Cleveland run No. 6. These six divisions are operated with 24 planes assigned by 15 pilots. Two of the pilots, George W. Hill and Frank Merrill, are assigned to Run No. 1, being detailed in Chicago. C. P. Olson is assigned to Run No. 2. He flies the route for two days and is relieved by the third by Fred A. Jones. Run No. 3 is flown for two days by F. E. Keltch, then Keltch is relieved for one day by Jones. Olson, Keltch and Jones fly in Kalamazoo.

Eddie Pontiac and C. W. March alternate on Run No. 4, the Bay City-Cleveland night run, both flying in Bay City. The regular pilot on the Chicago-Kalamazoo night run is Johnny Huber, who lives in Kalamazoo. Huber is relieved every fourth night by Hill. Hill on such nights flying the night end after having completed the Chicago-Muskegon round trip. After his arrival with the mail from Chicago, Huber ordinarily remains in Kalamazoo as a reserve pilot until the following evening at 5:35 central time when he ferries his Pioneer back to Chicago. By holding Huber in Kalamazoo until after the arrival of the company's planes from Muskegon, Bay City, and Pontiac, the company is assured that the mail collection in Chicago will be completed by planes from the Bay City and Pontiac plants will go through to Chicago on schedule, even though the total collection should prove too heavy for one plane to carry. Should this happen, of course, it would be a matter to the present light of things, the principal reason for holding Huber to relieve his services as a reserve pilot and start the incoming pilots land at the Kalamazoo Airport.

Run No. 6, the nightline line to Cleveland, is flown by three pilots, L. F. Bishop, "Cy" Caldwell, and A. H. Down, all of them residing in Cleveland and being off every third day. Eddie Merrill, in charge of the company's flying operations at the Cleveland Airport, is used as a reserve pilot on this run. Each of the three amphibians, incidentally, are taken all the way every third day, and kept at the Cleveland Airport for a thorough inspection.

ALTHOUGH, theoretically, the entire Thompson mail system operates on a definite schedule, happily the schedule can be made flexible enough to make practically all connections with other lines, yet serve their purpose within their own territory. The one exception to this is the mainline division which leaves Kalamazoo at 10 a.m., 12:30 noon, 3 and 5 p.m., daily, except Sunday. The daily service can be said to begin at 7:15 a.m. Central time, when the Chicago-Muskegon plane takes off from the Municipal Airport at Chicago after having collected mail from the several Michigan local plants. This plane arrives in Kalamazoo at 8:30 a.m. Central time, finding the Kalamazoo-Bay City and Kalamazoo-Pontiac planes wired up and ready to take off. At 8:55 a.m. all three planes leave Kalamazoo, Run No. 1,



Charles A. Robinson, general flight manager at the Cleveland Municipal Airport

being due in Muskegon at 9:35 a.m. Central time, Run No. 2 being due in Bay City at 12:05 p.m. Eastern time, and Run No. 3 due to arrive in Pontiac at 12 noon, Eastern time.

In the afternoon these three planes start from Muskegon, Bay City and Kalamazoo, respectively, all being timed to arrive in Kalamazoo at 5:35 p.m. Central time. In Kalamazoo all of Michigan's air mail is loaded aboard the Muskegon-Chicago plane, which according to schedule takes off at 5:40 p.m., arriving in Chicago at 7:30 p.m. So, generally speaking, operation of the company's daytime overland routes can be listed on a common picture of a ship's record in action, which, after having completed its loading and unloading operation, is reversed, depicting the vessel's movement in reverse order.

While this route is being flown, the company's amphibians are flying between Detroit and Cleveland, taking off four times daily from each city and connecting back 55 miles between the terminals.

The night run begins in Bay City at 7:55 Eastern time, when the Pioneer amphibian, used on this division, takes off for Cleveland, at which point it arrives at 11:30 p.m. Eastern time, in time to connect with both the east and west bound N.A.T. night planes. After picking up the Michigan mail from the east and west bound N.A.T. planes, the Pioneer, flown by the more pilot, departs at 3 a.m., arriving back in Bay City at 6:35 a.m.

Occasionally, though, the seasonal mail really starts with the can between Chicago and Kalamazoo. According to the schedule, Huber, loaded with mail collected from the Hoegs-Son Francisco route (due in Chicago at 6:40 p.m. Central time), is supposed to take to the air at 7:15 p.m., in order to beat the Detroit-bound train to Kalamazoo. It has been, however, previously arranged to hold this plane at the Chicago Airport until other mail planes are in, those later including Embury-Biddle and Interstate (due from Greenwald and Atlanta, respectively, at 7:10 p.m.), N.A.T. and Robinson (due from Dallas and St. Louis, respectively, at 7:20 p.m.), and Northwest Airways (due from Minneapolis at 7:30 p.m.). After taking aboard whatever Michigan and Canadian mail might be brought in on those planes, the Thompson plane can leave Chicago any time up to 10 o'clock and will beat the 5 o'clock train into Kalamazoo.

The entire Thompson mail schedule, in fact, is tightly flexible, either day or night. Michigan local planes being able to work in Chicago until a maximum of postage has been taken aboard—this delay being prolonged until danger of missing connections on the opposite end appears in the offing. One of the most important of

these connections was placed in effect on the day the writer began his flight over the company's mail lines. We speak of the Montreal-Detroit run, operated by Canadian Airways, Ltd. This service, daily except Sunday, begins in Montreal at 9:15 a.m. Eastern time, arriving at Ford Airport at 4 p.m. Eastern time, after stops at Toronto, Hamilton, London and Windsor. The Canadian plane arrives in Detroit and starts back to Montreal at 3:45 a.m., landing there at 11:30 a.m.

Postage figures furnished by Charles A. Robinson, general flight manager of the Thompson company, show that the Thompsons have carried 5,592 lb. of mail in July, 1958, 28,588 lb. in August, 10,897 lb. in September, 9,545 lb. in October, 9,280 in November, 8,706 in December, 8,402 in January, 1959, 8,486 in February, 9,300 in March, 14,783 in April, 15,595 in May, 17,941 in June, and 19,795 in July. The writer is cognizant of the fact that the August, 1958, jump was due to the Bus Motor Car Company, of Lansing, which representation signed a publicity stunt by sending 10 tons of air mail letters over to Thompson late in June. An especially marked amount of 17 Strouven-Detroiters was required to handle this shipment, which took place on August 21, 1958. Incidentally, that feat is used to represent the largest postage ever dispatched over any one company's lines on any one day in the history of American air mail.

The sudden increase in August, this year, no doubt was due to investigation of the Bay City-Cleveland division. That being as it may, the increase in postage from January, 1959, to the end of July, seems to indicate that the company served by the company is taking to the idea of making money by air, and suggest well for the operator's future success.

IN ADDITION to being contractors to carry the mail, the Thompson corporation is distributor in Michigan and Ohio for Stinson, Keystone-Leswing, Pricers and Gyge West planes. It is natural that their operating equipment should consist of such craft. Sixteen such planes are employed on Runs No. 2 and 3, over which just recently the company began to carry passengers. Runs Nos. 1, 4 and 5, however, are flown with open cockpit biplanes, all Pricers, except one Stinson used by Pilot Hill and the company uses a special aircraft for the Detroit-Cleveland overland service.

Run No. 6, was primarily designed as a passenger line, and, of course, employs Landing Amphibians. The mail carried over the marine division provides an incidental revenue, as do the passengers carried over the Kalamazoo-Bay City and Kalamazoo-Pontiac runs.

The Thompson maintenance staff is composed of 22 mechanics, 8 of them being in Kalamazoo, 4 in Chicago, 4 in Bay City, 2 in Pontiac and four others distributed between Saginaw, Flint, Detroit and Muskegon. Added to the ground personnel, of course, are the various terminal managers and their clerical help, G. H. Boushelle being located in Chicago, W. A. Goodyear in Kalamazoo, C. T. Taylor in Bay City and K. Matsuda in Pontiac. John L. Huber, operations manager, maintains headquarters in Kalamazoo, and Charles H. Sperry, Bay City, Cleveland division's chief manager.

In flying C.A.M. 37 Thompson pilots have a number of problems to contend with that practically no other mail operator has to face. Probably the major of these is a lack of adequate airports. The major portion of its stops being small cities like those of Michigan outside of Detroit, this is a handicap to be overcome. However, of these smaller cities there are several notable exceptions,

Kalamazoo, Pontiac, Battle Creek and Bay City having ports of sufficient size, excellently equipped. Pontiac probably has one of the finest airports in its section. Bay City, just behind the edge of that quietly populated upper sector of Michigan's lower peninsula, is building the James Cresswell Municipal Airport into something of which it can be proud. This port lies along the Saginaw River and can easily be made to include a base for air plants, flying boats and amphibians, converting Bay City into the departure point for air tourists bound for the vacation spots of upper Michigan.

Others of Michigan's airports, those at Lansing, Flint and Ann Arbor, for instance, are so poor Thompson pilots don't need to land during bad weather. It is apparent that under the new State Board of Aeronautics, operated by Governor Romney, the State is endeavoring the ownership of William B. Mayo and the directorship of Capt. Ray Collins, both of Detroit, this condition will be remedied. The State proposes to construct and operate a string of state-owned airports.

Airport personnel practice to question of the Thompson system is the nature of one stopping point is another. On the Bay City-Cleveland division, for instance, the pilot takes off at Bay City and flies but 12 miles to reach Saginaw. That is but 31 miles from Saginaw, and Pontiac is only 33 miles from Flint. Detroit 25 miles from Pontiac, and Flint 25 miles from Detroit, and Cleveland 85 miles from Toledo. Other divisions are similarly served, way points ranging from 12 to 200 miles apart. This, of course, has its advantages in the event of engine trouble, but in stormy weather and Michigan has its share of it) it means that the pilot must fly through it if the mail is to get through. Or jumps of several hundred miles, or more, of length, such as some of those on other mail routes, the pilot may, and often does, find it to his advantage to climb over, or go around a particularly mean area lying directly across his path.

With one of its divisions extending over, and two other extending around large bodies of water, the company's mail service has become a fact. Added to this is the ever-present pull of carrier through most of which the Thompson pilots must pass in getting into the Chicago Airport. Still another factor, although less undesirable, is the wind variations around lower Lake Michigan, due, it is said, to the separation of the lake.

Despite this, C.A.M. 37 recently closed its initial year of operation—a year during which the operation's planes flew 336,532 miles and had only two minor accidents. Total net results for the year were: Gross revenue, \$1,000,000.00. During this year, also, T.A.C. planes carried approximately 100,000 lb. of mail, with some being detached. Since completion of the first year the company's amphibian run, unfortunately, has not met with serious success.

According to Mr. Huber, the Thompson policy is to subject every new engine to a major overhaul after using it between 250 and 300 air-hours. Overhaul engines are provided for overhaul after not more than 200 air-hours. All of the company's major overhaul work is done at the Cleveland Airport, where the company maintains its principal shop, and it is distributed by Wright engines and engine parts.

T.A.C. is insured for potential liability, covering both passengers and planes, and endorsing fire, tornado, theft, piracy, liability and property damage, to the extent of \$4,000,000.

THE 1929 Schneider Trophy Contest

THE GREAT QUESTION here at Calicut, before the race is the maximum speed of the two new British contenders of the Schneider Trophy. While the secret is being carefully guarded, I can hazard a guess that if every anticipation is realized aircraft speeds may be raised to 300 m.p.h. over the standard three-hundred course. [Our correspondent's estimate on this point, written and mailed more than a week before the race, proved remarkably accurate, as the first trials of the Supermarine machine over the straightaway speed course on Sept. 10 showed 355.8 m.p.h.—Ed.]

The new Rolls-Royce engine in the Supermarine S-6

is, as one of the pilots expressed it, "a winded of an engine." At one moment it will tick down at 450 r.p.m. as gently as a car engine and the next be giving its full horsepower in a deep-throated roar. Yet throughout the whole range it will maintain an astonishing evenness of carburetion. On Aug. 30, the first Supermarine airplane of the two trials, with its new engine and increased cooling area for the water radiators, was given a final test, and this may now be regarded as mechanically ready for the race day.

The second British defender, the Gloster-Napier, has been damaged by accidents almost from the day it arrived, is work late, to miss four days of perfect flying weather. It is true that rough seas prevented any high speed machine leaving the slipway from the day of its arrival until Aug. 30, but as that morning the Gloster-Napier did raise its first real chance to get into the air on an extended flight simply because of a number of small but annoying problems connected, first, with getting a steady flow of fuel to the engine, and then of overcoming a fluctuating oil pressure. Later in the day the engineers ran the engine up on the beach. The Napier engine for the first time seemed completely happy, and all was ready for flight—but just half an hour too late to get another word before a rising wind.

SOME OF THE HITS of the race, and within areas of the navigability trials, only one British type is really ready. The Italian at last are here, but do not expect their racing machines to arrive for another day or so. The Italian team itself obviously had no part in the suggestions of withdrawal from the race and are generally glad to be here. The members tell me that their real estimate is placed in the two Macchi S.6s, the type on which Captain Macchi was killed; that the third and remaining Fiat is missing, a Macchi of 1927, and possibly two entirely central machines. One is a Savoia Marchetti, which is a central specific type with out-rippers to the tail unit to provide for a pusher drive by a rear engine. There is another engine in tandem driving a screw propeller, and the pilot sits between the two engines. Then there is the Puggio, a revolutionary conception described as a flying wing with, one guesses, some sort of radioactivity float with the pilot sitting in the wing and with one engine. How it can survive a choppy sea, use its aircrew without picking up green water and obtain control is,

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Both the Supermarine-Rolls Royce S-6 and the Gloster-Napier 6, in different directions, represent a departure from the policy of 1927. That summer frontal area was the outstanding feature, but this year the Supermarine designers have based their hopes upon a larger and more powerful engine, which in its standard Service form gives 830 hp., while the Gloster company has forsaken its biplane form for a monoplane, again at the dictate of higher power though in this case the frontal area of the power unit remains the same or is even a little smaller. Both engines are supercharged, and this, of course, has meant extra weight, greater length, and also more weight in the province of increased fuel.

The Supermarine monoplane with its very much more powerful engine, which burns something like 2 gal. per min. at full throttle, has definitely become a bigger machine all round, and finally broke the slender guise of the 1927 racer with the Napier Lion. On the other hand the Gloster machine has unquestionably gained in agility by the adoption of the monoplane form, seems to sit lower to the eye than the S-6 of 1927, and of the two, gives the impression of containing much more original design work in the direction of refining out only hard resistance but without interference between the components of the aircraft.

BEFORE SPEAKING of the position as calibrated here at Calicut before the race is this: Both machines have been designed to meet the same specifications, and with the specified maximum horsepower, theoretically there should not be more than 1 or 2 m.p.h. between them. Actually, assuming for the moment equal aerodynamic efficiency,

By Aviation's
BRITISH CORRESPONDENT



Front quarter view of the second landing Supermarine S-6 (Rolls-Royce)

the advantage must be with the monoplane which has the better power-weight ratio, and in this respect the Supermarine monoplane should have the best chance. Its Rolls Royce engine is only at the beginning of its development, whereas the Napier Lion must be reaching the limit down a power scale of view. The problem which is impending at all here is how much extra horsepower above that guaranteed under the original specification can the Rolls Royce racing engine safely be called upon to deliver for one hour at full throttle, and upon the answer may be said to depend the fight between the two machines for extreme speed. The Napier people with all their long experience of the Lion engine believe that whatever power they give above the maximum can be held without breakdown, whereas the Rolls Royce staff, we think, realize that they must exceed their contract to counterbalance the extra mass and weight of the engine which carries the engine.

IT IS AN OPEN SECRET that this new engine has not only performed extremely well, but has given a very much increased horsepower over that asked for, but, and it is a big but, too, no one knows yet how long the engine can be relied upon to run at that vastly increased horsepower before some trifling failure may set it out of commission. [Although no official figures have yet been given out, rumors of 1,400 or 1,500 hp. from the Rolls Royce were ripe before the race.—Ed.] Undoubtedly the Rolls Royce firm is in a better position in regard to supercharging, for it has a specialized knowledge of this method of increasing power and its superchargers are entirely the product of its own designing staff, whereas the Napier firm has taken the superchargers developed by the Bristol Aeroplane and Engine Company and with further co-operation from the Bristol-Thomson-Houston Company, British allies of the General Electric, has adapted it to the Lion type. Both superchargers, it must be said, are of the centrifugal fan type, and are placed



Visiting Officer Hughes receiving the official flag in the 1929 Schneider Contest

between the carburetors and the intake manifold, and appear to have been brought to a high pitch of efficiency eventually, while the running of the engines suggests that all carburetor difficulties have been overcome, especially in the Rolls-Royce unit.

Over more accurate equal power-weight ratios, visual comparison of the two airplanes suggested that the Gloster-Napier might be the faster because of a greater refinement in detail, but here again it was quite proposed to find first impressions justified by actual results. Unfortunately, however, the Gloster airplane plagues the eyes a bit. Both fuselages are made of duralumin upon the monocoque principle; but whereas the Supermarine is plated in parts very much on the lines of a ship's hull the Gloster fuselage has the absolutely smooth contour

of a yacht, and even on the floats there is not a protruding rivet to be found anywhere. Some figures given run as to the resistance of a soapline float with flush riveting, as compared with another in which ordinary rivets were used, show a surprising reduction in total drag.

In both British machines ordinary "stream-line" or line-tailor, bracing wires have been abandoned for even the theoretically correct and symmetrical, ailerons. These have been rafted to proper shape instead of being rolled, and I understand the difference between the two shapes is equivalent in the case of one of the soaplines to as much as a mph. This illustrates the importance of reducing all external forces of bracing to a minimum, and aligning with it the necessity of careful strapping for all the wires to enter the wings at right angles as nearly a right angle as possible. If be free view of the Gloster machine is examined from that standpoint, the reason for certain features will become more plain. The slight downward curvature at the wing roots is intended to provide better angles for the wires, and the position of the flow struts, which do not approach a crosswise point, is again dictated by the need of avoiding all possible interference.

The wing shape also has distinct significance, for it is an effort to combine the lower drag of a thin wing with the higher lift characteristics of a thick section. As the photograph shows, the wing thickens toward the outer periphery, with maximum thickness where the bracing wires are attached. It would appear that, apart from other advantages, such a wing should have the effect of increasing lateral control at low speeds, as the inner sections will begin to lift before the outer part. In plain form, also, the maximum chord occurs at the point of maximum thickness.

Both designers, too, have had in mind to enclose the pilot entirely by continuous underbrakes, the top line of the fuselage, leaving openings at the side only, with of course the usual curved type main wing struts. The adoption of this modification will rest with the pilots, and so far no decision has been made, though it would seem that another slight gain in speed should result. Another change which is certainly to be made is the fitting of different floats for the actual race, and this

after all is merely an effort to play for safety in the preliminary practice. Again the effect upon the extreme speed can only be awaited, but a few more miles per hour are expected. A number of the bracing wires being used on the Supermarine during the practice period are redundant, and they too will be removed for the race.

It is not necessary to describe the machines in detail, for the photographs are sufficient, but a few details of the observations in the water and of cooling systems will have a technical interest.

The Supermarine machine this year has adapted a new form of wing radiator, as the



The line of the Supermarine Sea Eagle

actual skin of the wing is a stressed member. It consists of two sheets of thin duralumin held apart by distance pieces and rivets to form a continuous waterway, and when the radiator was tried in the air it was quickly found that the cooling was not sufficient for the very much increased power being given by the engine. It was realized that the cooling area could be very much increased if the inner side of this radiator on the top and bottom of the wing could be used, and so three small openings were fixed to the underside of the wing tips, and at least double the number of smaller openings fixed to the top surface of the wing by the root to permit of a flow through the wing.

THE RESULT OF THE TESTS was not disclosed officially, but it appeared that a marked improvement was effected, and I believe that it is correct, broadly speaking, to say that the power which the new Rolls-Royce engine can deliver is governed now almost entirely by the efficiency of the radiator system. There have been suggestions that ethylene glycol should be tried for cooling, but the supermarine has not yet been ready. The problem of keeping the oil within reasonable working temperatures has thus far been much easier than in 1927, and no difficulty is being experienced as a result of the installation of a very elaborate system. The oil tank of the Rolls-Royce engine is now the hollow fin of the tail, and from here it is pumped to the engine along a thin lead which can be seen underneath the fuselage; so the return is a passive along triple deck of coolers running the

length of the fuselage. When it finally returns to the tank in the fin it is first squared against the walls of the upper part of the fin further to extract heat. Actually, it is calculated that the number of heat units absorbed by the oil, if it could be transferred into useful energy, would be equal to over 60 hp. The fuel, it will be remembered, was stored in one float in the 1927 machine, but with the supercharged engine fuel tanks of steel form the central portion of both the duralumin floats.

The engine of the S-E had an unpleasant habit of cutting out on a severe turn owing to the acceleration forces temporarily starving the carburetor of fuel, overcoming the action of the pump drawing fuel from the floats, and as these acceleration forces are even more intense on the new machine elaborate piping and reserve tank arrangements have had to be devised to insure a steady flow of gasoline under all conditions of maneuver likely to be required for the race.

MUCH TO EVERYONE'S SATISFACTION the Supermarine-Rolls-Royce flew perfectly the first time, and nothing was needed to correct a slight aberration in the incidence of the wings to get perfect trim. Control surfaces needed no modification whatever, and from that moment the sole concern of the designer and the pilots has been to control engine temperatures and adjust throttle controls to give the most suitable mixture. Everyone is impressed by the ease with which the Rolls engine starts, and the gentle way in which it will take over. This is in marked contrast to the Napier Lion, which has always been known when it is idling, and only begins to sing a sweet and even note when the engine is turning over quite fast.

So much practice was carried out on the first Supermarine machine that more work it had been returned to the works for a new engine to be fitted, and



Wing Officer Mathew, winner of the 1914 Schneider Trophy Contest

Owing to difficulties already mentioned, the Gloster-Napier until some days before the race had only been in the air for a minute. The racing trials were completely successful, but when the pilot, Squadron Leader Orlebar, closed to take the machine off, just as it got into the air the engine showed secondary signs of starvation. It is interesting to note that these warnings were given their first tests by the copulas of the British team, whose mark is equivalent to that of major in the U. S. Army, although he did not fly in the race itself. Squadron Leader Orlebar attained his rank at an age less than that of many first lieutenants in our Air Corps, the British promotion system being much more elastic



Front View of the Supermarine Sea Eagle. The last of the radiator is the triple deck of heat from the 60 hp engines

than our own—En f. I picked up again, but the pilot wisely decided to alight while he had good conditions. He flew gradually two miles, and as he returned reported that from a central point of view the Gloster was completely unsatisfactory. Since then, as has been explained, the engineers have been trying to tackle the installation troubles, which are complicated by the intensive piping system. The fuel is carried in the floats and the wings coat the radiator water, but the oil cooling is arranged differently to the Supermarine. The main oil tank is behind the pilot, and a complete section of the skin of the fuselage here is an oil radiator, with extension pipes to similar flat tube oil coolers on the top of the floats.

These may provide more cooling than is needed, but no ice is yet sure on that point, not on whether more radiator surface must be provided. It is hoped that the float radiators will be surplus and can be used for water.

THE WEEK BEFORE THE race had been filled with excitement and thrills, with narrow escapes from disaster to both teams. Seaplanes have been rescued in a waterlogged condition; there have been some anxious moments when the calm air was as rough as sea. Other types have perished and it seemed one more swim would be fatal, and last of all, one of the four hopes of the British team had such a narrow escape from collision with a large ship that it was towed in with its wing tip crumpled.

It was no credit to everyone when Friday morning dawned and for good or ill the choice of the British and Italian teams had to be announced. All the world knows now that the British had two of the new Supermarine Rolls Royce Seaplanes, and to the utter disappointment of the Gloster team had to rely for their third string on the Supermarine-Nagler Seaplane of 1939, the type which won at Venice.

The Italians chose two Macchi 67's, with four-cylinder engines of 1,000 to 1,200 hp, consisting of 38 cylinders arranged in three banks of six, and Major de Boreas' round bodied with a 1,000-hp Fiat engine and bigger floats to hold sufficient fuel for an hour's flying. One Macchi 67 had flown here for 15 min. only; the second had never been in the air when the inevitable crash had started, and the Italian team worked all night to repair a purchased float in the Macchi 52, which had nearly been sunk earlier in the day. Truly a hectic conclusion to a hectic week, for not until a last flight just before dusk could the British authorities decide whether or not the Gloster-Nagler seaplanes were to be included in the race.

The Gloster-Nagler seaplane has had a persistent and obscure trouble in causing a smooth flow of fuel to the

engine, which at first tended to cut out from starvation on terms due probably in part to acceleration forces, or an alteration in the air pressure on the air supply. This worried the subcontractor for days, and the Nagler designers then worked night and day devising the fuel system to ensure that a faster fuel tank, extended to feed the engine during these sharp turns, was always full. Then, this difficulty apparently overcome, Squadron Leader R. G. Orcher made a spectacular flight to late that as he took off the lights were aglow in the yacht club in the Italian and the navigation lamps in the shipping channel. Never has a high-speed flight been made under such poor light conditions.

Still the engine did not have a completely even fuel flow, but enough was learned to track the trouble, or so it was thought, and also to secure valuable data as to cooling areas needed for oil and water. The circular oil radiator which forms part of the skin of the fuselage behind the cockpit, plus the oil radiators on the float struts, had proved efficient, but more cooling was needed for the water. So as extra oil radiator was fixed each side of the cockpit, unfortunately spoiling the fine body lines, and the float radiators were switched over from oil to water. New sea-tang floats were fitted, and the final racing engines installed in both machines.

This occupied the Gloster team all the week. Interest, therefore, transferred to the Supermarine.

Rolls Royce seaplanes, and these two aircraft had crashed but endured a degree of damage hardly believable in what we are accustomed to regard as delicate high-speed aircraft only to be flown under perfect conditions.

The designers of this machine had broken new ground in three directions. First, no single-engine machine has ever had such high horsepower, and with this was combined a large gear reduction, a slow-moving and heavy propeller, and a much overbalanced cooling problem. The high horsepower introduced a very awkward problem of torque, the reaction tending to turn the starboard float when taking off. The 1937 racer overcame the much less pronounced torque by putting all the fuel in the other float and offsetting the machine slightly on the floats, but with a more powerful engine requiring a mileage of 100 gal this was out of the question, so originally the fuel was equally divided between the floats.

Very early it was apparent that torque was an important factor and the first time the S.6 was flown at full load disaster was very near. Already it had been found necessary provide greater cooling surface for the water radiators, and extra coolers—thin longer struts on each side of the two floats—had added to the overall weight. It was decided, too, to use a constant

aircrew and with this Flying Officer Waghorn made six unsuccessful attempts to take off porpoising so violently that he had each time to stop down.

GLASS STREAKS in the British camp, accentuated when it was seen that the seaplane was being towed in very low in the water and with the pilot sitting on one foot to balance the machine. Still she had kept aloft on a two-mile tow, and everyone was relieved when it was discovered a rubber washer in a float housing was bad and fissured and the starboard float had slipped over 300 lb. of water.

But the torque problem remained and new floats were built in which slightly more buoyancy was given one float and most fuel transferred to the other. An aircrew preflighting greater speed was used, allowing more power to be developed by the engine, and with this Flying Officer Aitcherley made a most successful flight, but a serious mishap occurred.

After being in the air for some time, the pilot felt a sudden difference in his controls, handed and to his amazement found both the starboard covering of the front starboard float stuck soon away, two of the petrol fuel pipes normally covered by the darning bent at one spot as if by a sudden blow, and even more startling, the edges of the broken darning fused like lead foil in a flame.

All the technicians held a consultation and it seemed that the most likely explanation was that the machine had struck in flight one of the many sea-walls always hovering about over the water here. The pilot had seen nothing and, of course, knew nothing of what had happened. The wonder was that a bird had not been struck before, as no bird could escape an object traveling at 100 mph. There still remained the hard nut, and at first it was thought the friction of two pieces of metal in that terrific air blast had generated the heat, or else violent oscillation of a flapping piece. The machine entirely wasted some delay and as opportunity was taken to install the final racing engine.

Meanwhile an experiment to overcome the way the Supermarine had of dipping her hull into the water and with a particular aircrew desiring to get on her

step, two small hydroplanes were fitted in the rear of each float to give greater lift on the water and assist the lift of the float to ride more on the water. They succeeded too well in one direction and again disaster occurred. The balance was completely upset for as Squadron Leader Orcher opened up the engine, the machine quickly began to porpoise, so violently that at times the big nose of the seaplane rose like a building house coming down with a resounding crash on the water, seemingly enough to shake the machine to pieces and crush the float struts into the engine directly above. These attempts proved that no help lay there and so soon as the machine was towed in, these extra hydroplanes were removed.

There only remained one solution to a problem in which the float area had been cut to the minimum necessary to allow the machine to get over the bump speed and on to the dry. A few pointed airfoils must be taken from the engine, even at the cost of putting extra stress on the power train. This was done and on the last day before the reliability trials, Flying Officer Aitcherley took out the seaplane, and had as severe a test as could be made, with a calm sea and no wind, he opened up the engine, got off at 45 sec. with full load, and all was well.

THE OTHER SUPERMARINE SEAPLANE had passed its tests with the same type of floats but without the hydroplanes the previous night. But on a straight run in Southampton Water, Flying Officer Waghorn for once made an indifferent landing, bounced like a rocket, crashed down in a seething of foam; porpoised violently and then drifted straight on toward a tug towing two enormous barges. Speed boats dashed to the rescue but the tide was rapidly erasing the seaplane, so so it seemed, into the water between the tug and the oncoming barge. The pilot perished on his 100 ft float, and the seaplane swam as it lay, but the high bluff bows of the barge passed on. Again disaster was averted at the cost of a dashed float and a crumpled wing tip. The delicate water radiators escaped and the damage was repaired in a night.

Great Britain had now two certain starters, so once again the Gloster-Nagler's troubles occupied attention.

Turning to the Italian effort, it must be said at once that in two cases, those of the little Fiat seaplanes and



Side-on view of a British Supermarine Tugboat-Catamaran, the Gloster-Nagler



Side view of an Italian Macchi 67, fitted powered with a 1,000 hp Fiat engine

Universal Seeking Reaction of Passengers

Smoke or Dust

Home Roasters Pilots

Highway Air-Making
System RecommendationContinental to Operate
Lakes-Gulf Service

Recess Seattle-Victoria Service

Air Mail Record

Set During August

The use of the highways in this manner is expected to provide a very helpful form of air marking, which is, of course, one of the simplest and most effective methods of aiding the important function of air navigation. The bulletin describing further details of this plan is to be released soon.

Arrange For Officers

The poundage figures are as follows:

Boiler	June	July	August
1	100	100	100
2	100	100	100
3	100	100	100
4	100	100	100
5	100	100	100
6	100	100	100
7	100	100	100
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91	100	100	100
92	100	100	100
93	100	100	100
94	100	100	100
95	100	100	100
96	100	100	100
97			

Country	1970	1975	1980	1985	1990
Argentina	171	2,048	2,048	2,048	2,048
Canada	171	1,021	1,021	1,021	1,021
France	171	1,021	1,021	1,021	1,021
Germany	171	1,021	1,021	1,021	1,021
Italy	171	1,021	1,021	1,021	1,021
Japan	171	1,021	1,021	1,021	1,021
Spain	171	1,021	1,021	1,021	1,021
Sweden	171	1,021	1,021	1,021	1,021
Switzerland	171	1,021	1,021	1,021	1,021
United Kingdom	171	1,021	1,021	1,021	1,021
United States	171	1,021	1,021	1,021	1,021
West Germany	171	1,021	1,021	1,021	1,021
Yugoslavia	171	1,021	1,021	1,021	1,021

Total ... 11,008 341,194 450,400 491,662

Flower Shipments

Flower Shipments

By Air Increasing

LOS ANGELES (caw) — After careful consideration of all factors re-

When the endpoints of G procedure request periods

Florida are amenable to the possibilities of the use of airways for the

Yellow Cab Changes Schedule

T.A.E. Using Two Cord

Air mail has been used for the shipment of exhibits to the great National Flower Show at San Leandro, California, from the East. Delivers sent from Washington, on the evening of Sept. 12 were expected to reach Oakland, the morning of Sept. 14, the opening day of the show.

WHAT OUR Readers Say

Parachute Struts

TO THE EDITOR:

Your editorial on "Pool Struts" in the issue of July 15, *Aviation*, is a subject which the Irving Air-Chrome Company is vitally interested in.

It is undeniably true that during the early stages of experimental work it was necessary for engineers to demonstrate the fact that it is possible to drop, by cut-throat or other form, from an airplane and still retain your arms and operate a parachute if the necessity arose. This was done several times on non-parachute planes, by Army and Navy divers.

It is thought, however, by the Irving Air-Chrome Company that entry should be made to drop such fool-hardy acts as swing how far one can actually drop out to ensure a record. There already have been several live tests, and the Irving Air-Chrome Company is absolutely opposed to this type of stunting. The usual personnel took in up with the Government officials when this was being done by the Army and Navy some time ago, and if I am correctly informed both the Army and Navy have a ruling that there should be no more delivery of this type of stunt.

It is the intention of the writer to take this matter up with the Department of Commerce to see if they will make a ruling to prevent this type of jumping.

George Warner,

President,
Irving Air-Chrome Company, Inc.,
Anchorage, New York

Light Plane Requirements

To Mr. Elmer

Understanding that the pages of your magazine have been thrown open to correspondence from subscribers, may I not add myself to the opportunity to express an opinion which I hope may stimulate some discussion among those interested in the maintenance and sale of small airplanes for individual use.

In the eyes of one who likes to think of himself as a professional pilot, it seems obvious that the article of placing too much emphasis on performance is all wrong. It is important only the manufacturers of small planes are so close to their market that they have failed to estimate the requirements of this market in their proper proportions.

and as a wing angle of descent, in of tremendous importance.

There are no traffic hold-ups in the air and the man who takes the most direct route and land near his destination consumes less traveling time than one who flies in a machine of brilliant performance which has to take off and land some distance from his starting point and destination.

Furthermore, if wing were made as important, the public would be rather shy in acquisition of low speed or slow than drive in a line of vehicles on a crowded highway.

I believe the *Aviation* Periodicals has a better understanding of the market for light planes than the aviation staff and it should be interested in an explanation as to why more manufacturers are not catering to the entry type, now being considered. Why is not research concentrated on the reduction of landing speeds, the use of slotted wings or their equivalents?

T. J. BURNHAM FOWLER, Jr.
(Anchorage, Ala.)

Editorial Comment FROM THE DAILY PRESS

Airplanes That Swim

THE manner of the waterborne emergency escape, who in fancy saw a picture that out of the sea is a tremendous height, and thus and through the air to fulfill rules of war, is facing a modified life without the flying boat, with the great danger air line as its most characteristic exponent, is increasingly evident.

When it is remembered that three-fourths of the world's surface is covered by ocean and seas, and that many centers of activity on the continental areas are located along rivers and lakes, the place that men seem to well as fly homes as a versatile factor in the air future.

John Coker's cockaded letter opened the way for the *Aviation* in making the Atlantic a connecting link between northern Europe and America, rather than a divider of the continents. Advancing beyond the oceanic vessel, the airplane has risen into the newer element, where speed is increased four times. Now the flying boat promises to give the sea a new role, that of a landing field, present all along the sea coast.

Recent seas across a new significance is clearly marked here for air traffic. Prior of high construction above the water level, they offer at the same time clear air highways, and landing fields on the flying boats may come to meet in the best of operation. New York, New Orleans and San

Francisco. When New York is projecting expensive landing fields on the sea of buildings, in order to have planes close to the city's center line is possible with the present airports, all demand more than an hour's drive the value of a flying ship that can land at many points in the Hudson or East River is obvious.

Aviation's first demand upon earth landing fields because, under present construction, unprofitable as carriers beyond a certain distance in size, aviation engineers have pointed out, since the necessity for rapid climbing to clear obstructions possible they landing areas causes the structural weights to increase in greater ratio than the carrying capacity flying boats, with practically unobtainable costs to take off and land, are increased in cost to a great degree, due to their more gradual climbing properties.

Explanation of wilderness regions becomes simpler through the flying boat, which can land on mountain lakes that could not be reached otherwise without enormous pioneering. Landing fields away from the flying boat's shorter, a short of eyes, glides and the amphibious, equipped with wheels, can also utilize any open ground for emergency landings. Airplanes that mean speed and safety, but which wait only for attention in the details, such as discovery of light metals which will start up under constant with less weight, for a dollar reduction.

CHRISTIAN SCIENCE MONITOR

THE BUYER'S LOG BOOK



G. E. Best Treating Furnace

ANNOUNCEMENT has been made by the General Electric Co., Schenectady, N. Y., of the development of a new type of electric furnace especially designed to heat treat aluminum alloys.

The new furnace is a pilot-type, cylindrical unit having two crucibles. One, rated 220 volts, 3 phase with a Y-connection, is in the top, while the other, in the lower side walls and bottom, is rated 220 volts, 3 phase with a Y-delta connection. At high heat the power consumption is 84 kw. and at low heat, 28 kw.

In a typical application of this furnace a run of castings weighing approximately 800 lb. plus a basket weight of 200 lb., both at room temperature, was placed in the furnace at a temperature of 960 deg. F. In 1½ hr. the center of the charge was at the best treating temperature, although for a third of this time the furnace was operating on low heat. A check showed that the outside of the charge next to the heating units was no higher than 560 deg. F.

In this typical furnace the working dimensions were a diameter of 4 ft. and depth of 6 ft. It was designed

Frishie Pitch Indicator

A NEW TYPE of pitch indicator has been placed on the market by Frishie Aircraft Products, New Haven, Conn. The principle of operation of the instrument is that of a pendulum geared with a ratio of four-to-one to a rotating drum. The instrument is actuated in a span aluminum case with a screw-on, clear-plastic-placed lens. Alighting screws also are aluminum plated.

Accurately fixed cup and pivot bearings of steel are used to mount the two bearing parts and require no lubrication. Damping liquid is used to overcome vibration effects and provide a steady reading under all conditions. The scale is easily readable and an adjustment is provided to compensate for variations of the instrument load.

The diameter of the case is 2½ in., the diameter of the lens, 3½ in. and the depth of the case, 3 in. The weight of the instrument is 8 oz. It is available with either glass or aluminum dial.

The diameter of the case is 2½ in., the diameter of the lens, 3½ in. and the depth of the case, 3 in. The weight of the instrument is 8 oz. It is available with either glass or aluminum dial.

Blount Tap Grinder

DEVELOPMENT of a new method for tap grinding has been announced by the J. G. Blount Company, Everett, Mass. The machine, driven by a specially designed 1½ hp. Westinghouse type SK motor is applied to the grinding of tap flats by the use of a taper arbor on the end of a spindle, using various sizes of grinding wheels.

For the grinding of small taps, wheels of a small diameter and width are used, while, on larger taps, grinding wheels up to six inches in diameter and 1 in. in width are employed.

These grinders are self-contained being driven by a special direct connected, fully reduced, belt driving motor with a speed range of 4,000 to 7,000 r.p.m. The motor is controlled by a field rheostat so that any grinding speed may be obtained up to 7,000 r.p.m. A smooth finish can be obtained by the inherently close speed regulation of the drive.



A charge being lowered into the General Electric pilot furnace.

to heat a charge weighing 1,000 lb. plus a basket weight of 200 lb. from 600 to 950 deg. F. in one hour, and to hold the charge at the latter temperature for 12 hr. The estimated maximum rate of the heating temperature is 17 kilowatt-hours per hour.

Oleo-Pneumatic Tail Wheel Unit

THE Cleveland Pneumatic Tool Co., manufacturers of Aero Oleo-Pneumatic landing units, have announced the development of a tail-wheel oleo-pneumatic unit. This offers advantages over the three-point suspension on this type of shock absorber which has been used extensively for landing gear.



Blount Tap Grinding Machine



SIDE SLIPS

By

Robert R. Osborn

MA J. A. of Buffalo, N. Y., was ordered into the office just now with the prize newspaper clipping of the week. It was from a small Virginia paper and concerned the recent flight of bombers from there to the West. "While Executive Presiders are enacting on the line at Langley Field ready to take off at a word's notice!"

R. H. D. Jr. reports a new flying thrill which is now available to the civilian flyer. He had been high-tailed for years by Army and Navy pilots who had a habit of looking him over with a condescending air as they wheeled by in their high-powered ships. Recently he was offered a cabin place with plenty of power in it out West and had the extreme pleasure of dipping his drive hat to the valiant and bearded hero in an Army two-seater.

With all of this winging and coming, going on to the Kansas, no later down would be a rivet of some sort before long. According to the latest on it at this time, Mr. R. H. D. Jr. was in it by W. C. B. of Baltimore, Md., the worm has turned at last. "Stearns Aircraft Group, Barn Center Corporation." Here's a company which was only refused to be taken over by a large holding group but up and buys the holding corporation.

Mr. C. P. McR. of Los Angeles, Calif., quotes from an editorial in a local paper. The great Denver De-Kurvis 12 powered engines and with make unnecessary forced landings under any but most extraordinary conditions.

We agree with Mr. McR. that necessary forced landings come often enough without complicating matters with unnecessary ones.

Mr. McR. also points out the possibility of aviation engines in the near future with relatively compact, according to a Los Angeles news item. "The ——— engine is being turned out on a schedule calling for one completed engine each day."

THE WALKING AMERICAN

AMERICANIZING ENEMY

POWELL is the *Lancaster Express* and *Illustrated* and *Illustrated* of *Lancaster* Wyo. In P. W. W. of *Lancaster* N. Y. — The gas jet before landing, checked, several thousand feet in altitude and circled around two or three times, and then touched down.

An interesting thought on refusing entrance flying is found in an item sent in by J. B. W.

"According to Maj. William B. Robertson of the Curtis-Robertson Company, spotting the flight both troubles have not bothered the flyers, although the American and the City of Cleveland were forced down by such causes. He attributed this to the use of forced landings."

Probably however would be paid for this, too.

Apparently the old tradition that an air race is always a big loss for the day in which it is held is still hanging on to our modern Ford time, according to the item from the *Star* John N. J. *Philadelphia Journal*, according to Mr. P. N. D. "Early in October several London entries are to have the opportunity of using the named Ford Liberty Tour."

In connection with the cross-country races in the Cleveland show a number of rearrangements have been made of forced landings caused by "foreign matters" is the gasoline tanks of several ships. It is our opinion that all foreign matters should be referred to the State Department and we are sure that a careful reading of the Kellogg Treaty will reveal some previous chapters for providing such forced landings in the future.

We see by the papers that one of the ocean-going liners get has been restricted radio apparatus in its ships, which will enable pilots to talk to the fields on route, and is considering carrying a telephone operator and a radioist so that passengers can be connected to long ground telephones or other ships. Here we observe not only possibility of a use for these refining industries

flights. Four hundred hours would be about the right length of time for the passenger to get his correct history.

There is another possibility to this airplane telephone business, which appears to us very much. Whatever we get into a traffic jam in the Airway (except we can't think of anything else) we can call the crash dinner until we are safe, sure. In the future there will be several radio jams, at certain, and with the radio telephone we will be able to call up the driver of the plane-moving airplane from a good old-fashioned wire and call him whenever the apparatus is used.

Another prize was given to W. F. D. of Stamford, Wt., being the description of an accident resulting from a landing on rough ground, as reported in the local paper. "It was apparent that the only safe landing that could be made under such poor conditions was a flip-flop."

A news item states that two residents living near an engine flying field have complained to the District Attorney that the field is a nuisance—"and that the powerlines hidden up in such that that they were obliged to close their windows. He said that the condition has been a nuisance for more than a year but that despite all of their efforts there had been little improvement." Our own opinion is that the residents didn't build it. They should close their doors, too, and if that doesn't keep the dust down on the field, they would have a right to complain.

The *Journal* *Aviation* came in a short time ago to make some conversation for a rebuilding of figures and statistics and concluded that a new airplane was constructed the other evening is one of the New Jersey airplanes and all attending pilots were required to dress correctly, which apparently meant top hats and evening dress. We said he was planning to travel a new landing gear on his Jersey very shortly and all persons who could obtain information to the event would also be required to dress correctly—decidedly Army branches. Another jack shirt and a sailor hat.



The Olden American Aeronautical Magazine

September 21, 1929

AERONAUTICAL ENGINEERING
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Patents Issued

AN APPROXIMATE EXPRESSION FOR

Airfoil Profile Drag

A Communication from
COL. V. E. CLARK

IN AN article "Design Your Own Airfoils," by the writer, published in the October 25, 1927, issue of AVIATION, data was furnished to enable the engineer to design an airfoil to meet the particular phases of his problem by giving proper span, depth, maximum lift and maximum drag for the angles of lift coefficient in which he was particularly interested. Recent wind tunnel tests on a model of an airfoil designed exactly according to the instructions given in this paper show quite interesting results. The model was quite roughly made and it is probable that a smooth model would have shown more favorable results.

The tests were made at an air speed of 40 m.p.h. and a density of one atmosphere. The aspect ratio was 7.5 and the mean thickness ratio was eleven. The center of median line was constant across the span with a maximum displacement from base line of .035 instead of .04. In other words referring to Table I all ordinates of median line were multiplied by the constant factor

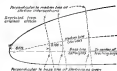


FIG. 1

.9. Thickness ordinates were multiplied by the mean by the factor 1.1.

At a lift coefficient of .001 (h, m, q, n, p) the L/D was 26.0. The maximum lift coefficient was .0035 and the center of pressure movement was extraordinarily

good. Judging from these results the qualities of this airfoil are sufficiently better than those of the "Y" to make it interesting. A number of check runs were made in the wind tunnel to be sure that there were no errors.

I have found that in making a study to determine the optimum properties of a wing for a specific design problem it would be inadvisable to have a simplified approximate expression for airfoil profile drag. There is a given combination of wing loading, aspect ratio, the airfoil thickness ratio at the root of the outboard which is best for any specific purpose. For example the increase in profile drag with increase in the airfoil thickness ratio should be weighed against the reduction in weight of wing spar chord members. An approximate expression for airfoil profile drag would save a tremendous amount of time in making a design study. With this in mind I have attempted to work up such an expression as follows.

Airfoil Profile Drag
Approximate Formula for Coefficient

$$K_{dp} = 0.00015 + 0.002 \left(\frac{T}{C} - 2 \frac{A}{C} \right)^2 + 100 \left(\frac{A}{C} \right)^2 + 8 \left(K_L - 0.02 \frac{A}{C} \right)^2$$

For coefficient of total airfoil drag (K_{dt}) should be added induced drag coefficient, $K_{di} = \frac{1.25 K_L^2}{R}$

Coefficients are for units: h, m, q, h, and m p.h.

$\frac{T}{C}$ = Weighted mean ratio of maximum thickness to airfoil section to chord length.

$\frac{A}{C}$ = Weighted mean ratio of maximum displacement of median line of section above straight line connecting leading and trailing edges—is chord length (DOH is a good compromise figure for wing).

K_L = Lift coefficient

R = Effective aspect ratio $\left(\frac{(KB)^2}{S} \right)$

Formula applies to best smooth fair airfoils of modern design—at full flight scale.

A portion of the article by Colonel Clark is the

Ordinate of Median Line, For Maximum Displacement .008 Chord Length

Distance from leading edge to ordinate	.025	.05	.075	.10	.15	.20	.30	.40	.50	.60	.70	.80	.90	1.00
Displacement in terms of chord length	.0024	.0041	.0049	.0051	.0057	.0063	.0068	.0072	.0074	.0077	.0079	.0080	.0081	.0082

Ordinate of Leading and Trailing Edges are 0

Table I

Expressed from original airfoil

MED-Thickness Ordinates, For Maximum Thickness .10 Chord Length

Distance from leading edge to ordinate	.025	.05	.075	.10	.15	.20	.30	.40	.50	.60	.70	.80	.90	1.00
Displacement in terms of chord length	.0239	.0016	.0152	.0194	.0415	.0416	.0495	.0472	.0426	.0359	.0304	.0268	.0237	.0209

Expressed from original airfoil

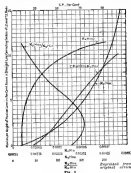


FIG. 2

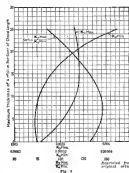


FIG. 3

October 17, 1927, issue of AVIATION is herewith reproduced: "An 'idealized' airfoil section upon which you base the data has been laid down. By superposition of this basic section—changing thickness or curvature of median line, or both, according to the methods outlined,—an infinite number of sections may be obtained which are ideal. Airfoils tapered in plan or in thickness ratio, or both, can also be constructed and their characteristics predicted. The thickness may be changed to meet the structural demands (wing beam depth, etc.) and the curvature of median line varied to obtain maximum lift coefficient or minimum drag coefficient to meet the performance requirements of a particular design problem. These latter two properties, characteristics,—important not only of themselves but also because they usually constitute an index of merit for all-round applicability,—have been 'predicted.' It takes a deal of ingenuity to venture such 'prediction' but after all, for the reason stated, these predictions of full scale characteristics probably will neither be confirmed nor contradicted. If they are so near accurate for the purpose of full flight performance calculation than low scale tested ones as judged by the National Advisory Committee tests, the adjustable airfoil series may be useful."

"The layout data for the basic section are given in Tables I and II.

"The layout method is shown in Fig. 1. A straight base 'line' is first drawn with the desired cord length. Perpendiculars to this line are erected at the stations indicated in Table I and the curved median line plotted and drawn. Then straight lines are drawn to each perpendicular to the median line as at point of intersection with the 'base' median line and the center of the section plotted from Table II and drawn.

"If a larger maximum lift coefficient than that for the basic section is desired, at the sacrifice of drag and center of pressure travel, the ordinates from Table I are multiplied by a constant factor greater than unity to give the desired characteristics predicted in the chart, Fig. 2. Conversely if low drag or center of pressure travel is more important than high lift for the particular use the revision factor may be less than unity.

"If deeper square than oval is contained within the basic section are required for strength and rigidity at the surface of drag, the ordinates from Table II are multiplied by a constant factor greater than unity, a 'thicker' section drawn around the median line selected and its characteristics predicted from Fig. 3.

"Combinations of camber and thickness are infinite."

NOTES ON THE DESIGN OF THE

N.A.C.A. Cowling

By WILLIAM H. McAVOY

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THE N.A.C.A. type cowling has been applied to many types of aircraft using radial air-cooled engines, with varied success. In some installations, the modifications which were made to the N.A.C.A. cowling resulted in lowering the cooling efficiency and in improper engine cooling. It is the purpose of this article to present some additional notes on the design of the cowling, calling particular attention to the fact that the nature of the problem is such that, at the present time, no set rules can be followed. It is merely due to the differences in the detail design of various engines. However, there are certain definite facts which influence the cowling design and must be considered, judgment being exercised in the evaluation of the importance of these facts.

The air which flows over the fuselage of an airplane with an enclosed radial air-cooled engine installation,

is quite turbulent because its flow has been disturbed by the irregularity of the engine section. The main idea of the N.A.C.A. cowling is to keep a smooth flow of air over the fuselage and at the same time properly cool the engine. The result is an apparent decrease in the drag of the airplane. The general layout of the N.A.C.A. cowling is shown in Fig. 1.

The drag of a fuselage is increased by any projection on its surface which disturbs the smooth flow of air over it, and the drag of a fuselage with an air-cooled engine may be reduced and made to approach more nearly the drag

of a perfect streamlined body, if the engine is completely covered. Also, the larger the fuselage behind an air-cooled radial air-cooled engine, the less is the increased drag due to the engine. From the standpoint of drag, there is no reason to make the maximum diameter of the fuselage smaller than that of the engine. Cowling is further justified in this connection by the fact that it is necessary for proper cooling, since an uncowled radial air-cooled engine can lose air when placed in front of a large fuselage. This is also true if the engine is incorrectly placed with respect to the body behind it, in which case some cylinders will be cooler than others.

It is reasonable to assume that if an air-cooled engine will cool without cowling, then by the use of it and the control of the airflow which it allows, there should be no real difficulty in maintaining the same degree of cooling so fast it should be better.

Before dropping the N.A.C.A. type cowling to properly cool an engine, it is necessary to know the engine operating temperatures and the characteristics of the engine. However, in such case with a general knowledge of the allowances which should be made by local changes in the cowling to care for such irregularities of the engine and installation in various aspects, most fully suggested, valve housing, shape of heads, and local projections behind the cylinders which would hinder the free out-flow of cooling air, etc., a cowling can be designed which will cool and give an increase in speed and be in full accordance with our developed

in the more costly way using thermocouples. But it must be remembered that overheating of an air-cooled engine is often difficult to detect before damage is done, so it does not always quickly or readily object to the way in which it is installed and may run for quite a long period before adequate cooling of some part makes itself felt in some form, such as pitted rings on an individual cylinder. Many operators of air-cooled engines assume that the oil temperature indicates engine temperature to the same extent as the water temperature on water-cooled engines does, which is not true. Since the only indication which a pilot has of overheating is either the loss of power, which may be too gradual to notice until damage is done, or an increase in oil temperature, which is less affected by the state of the head. The only sure way to get the best cooling, and one which meets the manufacturers' temperature specifications, is to follow a definite schedule for measuring engine temperatures. By closely studying the data thus obtained, the cowling may be modified where necessary. Thermocouples should be placed at as many points on the cylinders as possible, paying special attention to the head, and not using any one point as a criterion if representative indications are desired.

Giving due consideration to the foregoing, the actual design of the cowling may be started at the case. The



FIG. 1

shape of the nose of the outer cowling is determined by the diameter of the engine; the distance the propeller is forward of the cylinders; a consideration of the flow of air to the engine parts to be cooled; a good entry to the nose to increase the cooling air from that which flows over the fuselage; the shape and size of the body behind the engine to which the cowling must be fitted; and a realization that, since it has little effect on speed, there is nothing so to be gained in keeping the opening at the nose small to the point where it becomes difficult to get an easy flow with a sufficient amount of air to the cylinder heads. The quantity should be controlled by the size of the exit. A gain in propulsive efficiency is obtained by the use of the cowling, and it is believed to be entirely due to the smooth flow of the air after it leaves the propeller and passes over the outer surface of the outer cowling near the nose.

To locate the point from which to lay out the shape of the nose of the outer cowling, past experience and a study of conventional engines indicate the following rule, which is illustrated in Fig. 2. Through the center line of the cylinders at their maximum diameter draw a line (1) forward to the propeller plane and which, if extended, would intersect the outer line of the crankshaft ahead of the propeller at an angle of 30 deg. Allow-

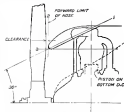


FIG. 2

ing clearance, the line (2) may be drawn perpendicular to the crankshaft showing the limit at the forward position of the nose. Through the intersection of these two lines (1) and (2), draw another (3) parallel to the crankshaft. There are now two lines, (2) and (3), (2) limiting the forward portion and (3) the size of opening in the nose of the cowling. The curve of the nose may now be drawn tangent to these two lines, as is shown by the accompanying sketch. The rear line of the outer cowling should be so designed as to allow the cooling air free passage over the top of the cylinder head.

The inner cowling is, in effect, a deflector. It is very important and must not be omitted. The inner cowling and the inner fins of the outer cowling together with vertical deflectors placed between the cylinders, are used to direct the airflow where needed. The air passage through the cowling, from the nose to the rear, should permit an easy flow, directing the air as little as possible from the direction of flight and still directing it to the hot parts of the cylinder in fairly parallel to the cooling fins as is preferable. The point of intersection of the lines of the inner cowling with the outer line of the cylinder should be at about the top of the piston when the latter is on bottom dead center, and from this point the cowling is faired both into the nose at the crankshaft, and into the opening at the rear. The vertical deflectors share with the rear opening in controlling the quantity of airflow and should be considered together. The deflectors should be designed and placed to make off the air which flows into the nose and work in a curve and not allow it to flow freely through the spaces between the cylinders without coming in contact with them. Figure 3 gives an idea of the location of the vertical deflectors. The success of the deflectors is due to their being less air to better advantage than can be done without them, and thus allowing the opening at the rear to be reduced. It might be well to mention here that, since the deflectors determine the path of the air and the exit opening determines the quantity, by properly proportioning and changing the opening, variations in cooling may be obtained for unusual flight conditions and the engine operating temperatures kept fairly constant regardless of air temperatures. Thus, the generally known advantages of the air-cooled engine in countries of climate which have been borne out from actual experience are enhanced.

by the use of the crawling. So far, nothing has been said of the crankcase other than that it is completely oiled. The crankcase has to deal with a certain portion of the cylinder heat which it gets from the oil, the amount depending on whether the cylinder heads are

which the cooling air re-enters the outside air. It is believed that it should do so nearly tangentially with a reasonable length of opening, but further work is necessary to accurately fix the limits of the angle of re-entry and the size of the opening. It is also indefinite as to whether the nose cooling lines extended should turn into the fuselage or extend outside and parallel to it, as shown by Figure 4.

The coupling has other possible advantages, some of which should make it attractive to engine designers. There are details such as overhead valve gear, for instance, which are limited in their compositions of design for better efficiency because of their increased drag, and which are no longer a problem when the engine is completely cooled.

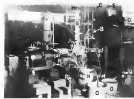
The cowling may be used as a carburetor air heater, depending upon whether the intake air scoop draws air from the outside of the cowling or from the inside where it is heated by having passed over the cylinders. Any air used from the inside of the cowling decreases the size of the tank with its known resultant advantages. With the scoop inside the cowling, there is less possibility



piston to be turned was placed within a special steel cylinder, constructed of stainless steel taking turned and ground to the correct internal diameter. The cylinder was made in two sections, one section extending from the head of the piston to a point just below the lowest fuel ring, and the second covering the rest of the piston skirt. An asbestos board filler plate was fixed between the two parts, providing an air gap of $\frac{1}{8}$ in. on the inside surface of the cylinder. Each section was fitted with a separate water jacket and insulated thermally from the other as thoroughly as the available space would permit. The two sections were held between end plates of asbestos by through bolts and the whole assembly lagged with magnesium cement to a minimum thickness of one inch. One end of this cylinder, that corresponding to the head end of the piston, was open for connection to the bearing seat while the other end was closed by an asbestos plate in which passages for circulation of air were provided.

Heat Furnished by Burners

The mixing coil was placed over a furnace with the head of the piston down. Heat was supplied by means of two Meker burners using gas from the city mains. The burners were so mounted in the furnace that the flames converged toward the center of the piston head. The furnace, of plain steel, was built up without mortar to resemble a short section of a small diameter with a square top opening whose width slightly exceeded the diameter of the piston. This opening was covered



A photograph of the apparatus used in the tests. A—Piston; B—Water jacket; C—Thermocouple; D—Thermocouple; E—Thermocouple; F—Water jacket; G—Water jacket; H—Water jacket; I—Water jacket; J—Water jacket; K—Water jacket; L—Water jacket; M—Water jacket; N—Water jacket; O—Water jacket; P—Water jacket; Q—Water jacket; R—Water jacket; S—Water jacket.

with $\frac{1}{8}$ in. layers of asbestos board containing concentric circular holes slightly smaller in diameter than the piston. Air entered the furnace through an opening at the bottom of the front side and the gases were vented through the similar opening at the top of the rear side, under a metal screen shielded on top by a sheet of asbestos. A small electric fan provided a draft through the furnace in the desired direction.

Cooling water entered at the bottom of the jacket of each cylinder section and passed around it, to leave at the top. The temperature of the water was measured as it entered each jacket and again as it left. The amount of water passing through each jacket was controlled by a needle valve, and the exit water was collected in two separate buckets on separate scales. The water was supplied by gravity from a constant head tank in which it was kept at 160 deg. F. by having low pressure steam flow through it.

Crankcase Gas Cooling System

The cooling air to simulate the cooling effect of the crankcase gas was drawn through holes in the top plate of the coil and sent through a cross-hatch pipe, which cleared the bottom of the piston about one inch, to the intake of a Bush-Ryan vane type air pump, driven by an electric motor. The delivery of the pump was proved through an orifice hole, the pressure in which was read by a manometer. Control of the air flow was obtained by bypassing the delivery of the pump back to the intake through a three-quarter inch pipe. The pipe leading out of the cylinder was lagged for the first two and one-half feet of its length, at the end of which a tee provided vent to a thermometer. The distance from the piston to the thermometer reduced the error due to direct radiation from the piston to the thermometer.

The temperatures in the piston were measured by means of eleven true-constant thermocouples located as shown in the plans. Each wire was inserted in a hole drilled in the piston with a No. 39 drill, the end of the wire inserted in the hole and the metal pressed back into the hole to grip the wire. The two holes for each pair of wires

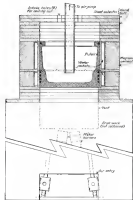


Fig. 1

forming a thermocouple were located not more than $\frac{1}{8}$ in. apart.

In both pistons, wires of the thermocouples had to be gripped from the inside and come from the outside of the piston due to mechanical limitations. It was intended that each wire make its contact at the inside surface and while this probably did not occur in all cases, for the sake of uniformity it was assumed to have done so and is so shown on the plots (Figs. 3 and 4). The error so introduced would be, at a maximum, equal to the comparatively small temperature gradient through the piston wall.

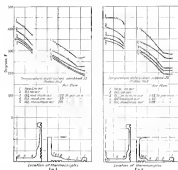
Readings of temperature were taken with a Leeds and Northrup potentiometer provided with an automatic cold junction and calibrated to read directly in degrees Fahrenheit.

Two pistons were tested No. 1, Fig. 5, from the Wright J-43 engine, a one piece casting of aluminum alloy (99 per cent Al, 92 per cent Cu) furnished with four rings above the wrist pin and one at the bottom of the skirt, and No. 2, Figs. 6, 7 and 8, from the Wright J-5 engine, a one piece casting of the British "V" alloy, with two sealed ribs extending to the bottom of the skirt and two smaller ribs meeting the inside end of the wrist pin below all 14 planes perpendicular to the axis of the wrist pin. This piston carried three rings above the wrist pin, the lowest of these being an oil scraper with three holes drilled in a deep oil groove in the piston wall just below the ring, and one ring at the bottom of the skirt. Shallow circumferential oil grooves were turned in the piston skirt.

The sectional views of these pistons, Figs. 5 to 8 inclusive, are drawn to three-quarter scale, but for clearness and simplicity are not dimensioned the purpose being to show the relative proportions of the various parts rather than their exact sizes.

Method of Testing Tests

A series of five tests was made on each of the two



pistons that were used in conducting these experiments.

- (1) Without oil on piston and without cooling air circulation.
- (2) With oil and without air.
- (3) With oil and maximum air flow.
- (4) With oil and maximum air flow.
- (5) With oil and maximum air flow.

The variation in the air flow was obtained by means of the by-pass control previously mentioned. "Maximum" air flow occurred with the by-pass completely shut and "minimum" air flow with the by-pass completely shut and "medium" flow at a pressure in the orifice hole approximately half way between the extreme pressures.

Piston Temperatures

The temperatures, read by means of the thermocouples, on plotted against thermocouple location, in Figs. 3 and 4 for Piston Nos. 1 and 2 respectively, for the various conditions of testing. Due to the error of thermocouple contact location discussed elsewhere in this report, these points cannot be considered exactly at the piston surface.



Left—Water jacket assembly showing thermocouples, water jacket and top cover. Right—Water jacket assembly showing lagging and base on which it was mounted.

The curves are valuable only as a general indication of the temperature conditions to be found in the pistons. The comparatively large drop in temperature upon the exit of oil on the cylinder walls, is accounted for by the fact that the heat conductivity of oil at the temperature of the test is approximately seven times that of air. This shows the cooling system to work more efficiently by providing a better medium of heat transfer between the piston and cylinder wall. It was anticipated that, due to the absence of piston motion during the test, the oil would collect by gravity in the ring belt. That this actually occurred is shown by the fact that the largest temperature difference between the cold and dry condition occurred at the ring belt where the oil completely filled the clearance.

In an operating engine, the distribution of the oil



Figure 8. Piston No. 1 with thermocouple leads. Distribution of leads on top of skirt (Fig. 8).



Figure 9. Piston No. 2 with thermocouple leads. Distribution of leads on top of skirt (Fig. 9).

would be more uniform, though it is doubtful if there would be enough oil to completely fill the clearance. Therefore, this effect might be diminished and the resultant would probably be somewhere between the conditions with and without oil.

The effect of the ribs in Piston No. 2 seems to be a flattening of the heat temperature curve and a steepening of the skirt curve. Slightly higher temperatures were obtained with the ribbed piston for all conditions except condition No. 3, with oil and maximum air flow. This piston is of lighter section throughout than No. 1 and the

direct contribution of the ribs to cooling becomes apparent only with the use of the internal air blast. The smaller number of ribs and smaller quantity of material in the ring belt are causes contributing to the higher temperatures found in the ring belt of the ribbed piston. The determination of the relative effect of the smaller sections and of the ribs has beyond the scope of these tests.

Distribution of Heat Removed

A plot of Btu's per minute removed through rings, skirt and air blast is given in Fig. 9. It is evident from the decrease in piston temperature that the addition of oil to the cylinder walls must have increased the heat flow through the piston. The fact that this was not indicated in the case of piston No. 2 (ribbed) is undoubtedly due to experimental error. Possibly the rate of gas flow to the burner may have varied. It could be desirable to construct the heating element of the apparatus so that, within reasonable limits, the temperature at the center of the piston head could be held constant regardless of the method of cooling employed.

In Fig. 10, the heat dissipated through the various elements, is plotted on a percentage basis. This shows most clearly the effect of the various rates of air flow and the practically negligible effect of the oil as regards changing the distribution of heat removal between rings and skirt.

Probable Errors

Thermocouples were not calibrated against a standard as their only purpose was to indicate the relative temperatures existing in the various parts of the piston. Their exact point of contact could not be definitely ascertained, as was stated above, so that calibration would have fallen within the experimental error.

The error in timing the tipping of the beam on the water scales was estimated to be one second at a maximum, giving a probable error of from 0.8 per cent to 3.5 per cent in this reading.

Thermometers were calibrated and found to be correct within ± 1 deg. F. and the mean value of several readings was used in the tables.

Radiation loss through the lagging was determined by experiment to be 3 per cent of the total heat dissipated through the jacket. This error should have remained approximately constant for all runs, and therefore has had little effect on the comparative value of the results.

The error in air flow measurement was small, as a calibrated orifice was used and correction made for non-Newtonian and laminar pressure.

Other errors, such as radiation to thermometers, variation in gas flow to the burner and miscellaneous heat losses probably accounted for an appreciable error



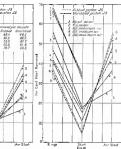
Left—Piston with thermocouple leads (Fig. 8). Middle—Close-up view of piston (Fig. 8) in test cell. Right—Piston in test cell with method of mounting burner to provide cooling effect.

in the measurements of total heat flow. The error introduced by the removal of all the internal gas motion of the curves No. 1 and No. 2 in Fig. 9 is 20 per cent at its maximum. Percentage distribution of heat flow as given in Fig. 10, however, is a much more reliable figure than the absolute quantities, and the errors in Fig. 10 are believed to be small.

Conclusions

Within the limited range of temperatures and with the two types of aluminum alloy piston used the following conclusions may be made:

- Without any air flow inside the piston between 65 per cent and 68 per cent of the heat dissipated goes out through the ring belt and 32 per cent to 35 per cent through the skirt.
- The forced flow of fresh air over the inside of the piston resulted in a lowering of these figures from 35 per cent to 56 per cent at the ring belt and 44 per cent to 18 per cent at the skirt, depending on the rate of air flow, the lower figures corresponding to the maximum air flow in each case.
- The forced air flow carried away from 27 per cent to 55.5 per cent of the total heat dissipated from the pistons, depending on the amount of air flow and the design of the piston the ribbed piston dissipating 43.5 per cent to 55.5 per cent and the unribbed one dissipating 27 per cent to 46 per cent to the air.
- The presence of a film of oil on the cylinder walls has no great effect on the relative amounts of heat



dissipated through the ring belt and through the skirt.

5. The increase, due to air flow, in the total amount of heat flowing through the pistons was from 25 per cent to 55 per cent of the original amount, depending on the rate of air flow. This increase was approximately greater in the case of the ribbed piston.

Further considerations with this apparatus are planned, with a view toward comparing pistons with wider extensions in design. An attempt will be made to measure the cooling effect of the crankcase gases in an actual engine, and this result will be used as a criterion to establish the rate of air flow in this apparatus so that it will approximate engine conditions.



TECHNICAL REVIEWS



NACA Technical Note No. 316 Wind Tunnel Tests On a Model of a Monoplane Wing With Flapping Airfoils, by Montgomery Knight and Edward J. Swisher

This report describes preliminary wind tunnel tests on a model of a monoplane wing equipped with wing tip flapping airfoils. Lift and drag, as well as rolling and yawing moments, were measured.

The rolling moments were roughly independent of angle of attack and the yawing moments were small. With the airfoils neutral the maximum drag was more than twice that of the wing without airfoils.

NACA Technical Report No. 311, Aerodynamic Theory and Test of Strut Forms, by R. H. South

The whole study under this title is in two parts, only the first of which is reported here. In this part the symmetrical curved flow about an empirical strut of high sweep merit is found by both the Blasius and the Joukowski methods. The results can be made to agree as closely as desired. Theoretical stress surfaces

as well as surfaces of constant speed and pressure in the flow about the strut are found. The surfaces pressure is computed from the two theories agree well with the measured pressure on the fore part of the model but not so well on the after part. From the theoretical flow speed the surface friction is computed by an empirical formula. The drag is computed from the friction and moment pressure closely equals the whole measured drag. As the pressure drag and the whole drag are accurately determined, the friction formula also appears trustworthy for such flow shapes.

The Glean of Wood, by T. R. Truax, Senior Wood Technologist, Forest Products Laboratory, Bureau of Research, Forest Service

The purpose of this publication is to bring together material information about gleaning and gleaning, to set forth important principles of control in the gleaning operation, and to outline methods that have been found to give satisfactory results.

ANALYSIS OF THE *Wing* AND OTHER *Indeterminate Structures*

By JEAN FRADIN and ARMAND THIERLOT

(Continued from page 643)

PART IV

IN THE last article we gave a method for computing the loads in the spars of a wing considered as an indeterminate structure. We calculated the reactions of the compression members on the spars and found the true value of their bending moment. We will now find the calculation of this conventional type of wing structure by determining the loads in the compression members.

The problem which consists of analyzing a wing structure considered as an indeterminate system is the general case, i.e., when the external forces acting on the system are a resultant force passing through the elastic center and a torsional moment—is a problem without a general solution. On the contrary, a particular method must be used for each particular case.

Nevertheless, the reaction of the spars through the compression members depends upon the rigidity of the spars in torsion, it is obvious that such spar does not remain parallel with itself as it deforms, but has a tendency to rotate and the magnitude of this movement will affect the deflection of the spars as well as the

moment in the compression members are a function of the torsional rigidity of the spar and of the elasticity of the compression member.

But the spar cross-sections generally adopted (regular shape and often not homogeneous) do not permit the



Fig. 3

composition of the value of the torsional deflection of the spars; and therefore the true magnitude of the bending moment in the compression members.

Hence it can be seen that the calculations of the bending moment in the spars made according to the method outlined in the last article are obviously on the conservative side. It was shown that, if the spars were calculated in accordance with standard practice in such a manner that their margin of safety is equal to zero, actually in most cases they have a positive margin.

The calculation of the compression members neglecting the torsional deflection of the spars is also conservative since we suppose that the moments M (Fig. 4) are entirely absorbed by bending in the drag member instead of being relieved by the spars in torsion.

But the rigidity of the spars in torsion is generally so much greater than the rigidity of the compression members in bending that the following calculations, based on the above assumption, give the loads in these members with reasonable accuracy.

For the same reason in the last article the presence of the compression members was not taken into account in

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writing the equation of least work, and the spars were considered alone.

We will demonstrate that if the factor due to the torsional moment had been taken into consideration in the equation of least work, it would only have proved that the magnitude of the moment M is such that it balances the vertical reactions Z of the compression members on the spars.

Let us call T the total work of the entire system: spars and compression members. The derivative of T with respect to the torsional moment for the spars equals zero since we have neglected the torsional deflection of the spar. We have for the compression members:

$$T_1 = \sum_{i=1}^n \int_0^L \frac{1}{2EI} dx$$

where n is the number of compression members, L is the length of the compression members and E the bending moment in the compression members.

The expression for the bending moment in the compression member is:

$$M_1 = Zx - M$$

therefore we have:

$$T_1 = \int_0^L \frac{(Zx - M)^2}{2EI} dx$$

whence:

$$\frac{\delta T_1}{\delta Z} = - \int_0^L \frac{Zx - M}{EI} dx$$

whence:

$$- \frac{ZL^2}{2EI} + \frac{ML}{EI} = 0$$

and:

$$M = \frac{ZL}{2}$$

which proves that the moment due to the fixed end, M , at each spar is equal to the moment of the vertical reaction Z acting at the other end of the compression member.

Therefore the vertical reactions, that we have calculated in the previous article, permitting the composition of the moments M give all the elements necessary to compute the loads in the compression members. These members should be treated as beams in bending and compression due to the drag loads.

Analysis of a Wing with a Rigid Covering

In a wing having a rigid covering, the compression members are generally curved, consequently the loads acting on the spars may be determined by a consideration of elastic center as previously outlined. The torsional moment is taken partly by the spars and partly by the rigid covering. The problem consisted of finding the torsional stress acting in the covering for a given torsional moment. That is, in most cases, the torsional moment given by the nose drive condition. This moment is found by multiplying the shearing stresses on front and rear spars by their respective distances to the drive center and adding the results. This is done in the same manner as for compression members, for a certain num-

ber of stations taken along the spars, beginning at the wing tip. The increase of torsional moment is calculated for the station considered and added to that of the former station.

The problem considered as a whole, that is the wing structure taken as a line with fixed beams, the walls being



Fig. 4

microsegment by the ribs (Fig. 4) and submitted to any loading, does not yet admit of a practical solution. However, a good approximation might be found by a mathematical integration of the results of tests.

The method for computing the stresses in such a wing structure, given below is based solely upon the fact that the results obtained are closely enough in agreement with the experimental data. With a plywood covering calculated for a shearing stress of 800 to 1,000 lb. per sq. in., it has been found that the compression or tension stresses are not critical and furthermore that they relieve the spars of a certain part of their load. In the low angle of attack loading condition this relieving factor is generally about 30 per cent.

For the other conditions this factor varies as the ratio of the torsional moments, i.e. as the product of the resultant airload by the distance from the center of pressure to the elastic center. The shearing stress in the wing covering due to the torsional moment can be calculated with a sufficient degree of accuracy by means of St-Venant's formula:

$$= \frac{M_1}{2GJ}$$

Where J is the polar cross-sectional area and G the wall thickness. This stress added to the shearing stress due to the chord load will give the total shearing stress in the covering.

The numerical calculations are carried out in the same manner as those given in the previous analysis.



Fig. 5

First the spar is considered alone and its cross-section compressed for a two margin of safety. Second the plywood covering is calculated for the shearing forces and torsional moments as indicated above.

It appears that the spars of a wing calculated by this method will have a positive margin between 15 per cent and 25 per cent.

(The authors are indebted to the Fokker Aircraft Corp. for the use of the above example.)

bending moment in the compression members—as shown in Figures 1, 2 and 3.

Fig. 1 shows the undeformed basic structure formed by the spars S_1 , S_2 and the compression member M .

Fig. 2 is a schematic picture of the same structure when the spars have deflected in such a manner that they remain parallel with their original position.

Fig. 3 gives a true view of the system after deflection. The resultant deflection of the spar and the bending

✶ PATENTS ISSUED ✶

Patent No. 1,726,435. *Seaplane Float.* Boris V. Kozlov-Kozlovsky, New York, N. Y., assignor to Edo Aircraft Corp., College Point, L. I., N. Y. Five claims.

A seaplane float of the V-buoy class having a transverse strip cut far from the center of gravity, each side of the bottom being shaped with a few large scallops which extend longitudinally from a region near the stern rearward to the transverse strip, said scallops exhibiting in cross section a series of broad and shallow inward curves extending between the keel and the chine and separated from each other by substantially sharp or narrow ridges, the scallops being adapted to throw water sideways and downward with consequent formation of air gaps in the scallops as the result of water breaking off the ridges at high planing speed, whereby to reduce frictional resistance during take-off the slope of the scallops outward away from the ridges being such as to keep the water from breaking off the ridges at low speed while the float is getting on the step.

Patent No. 1,726,118. *Method of and Means for Controlling Airplanes.* Frederick Handley Page, London, England, assignor to Handley Page, Limited, London, England. Seven claims.

In laterally controlling seaplanes flying machines, herring wings projecting laterally from opposite sides of a central fuselage, said wings having forewardly located secondary wings separated from the main wings to produce through slots, means for regulating the flow of air passing through said slots in said wings, elements pivotally connected at the rear of said main wings, and means in operative connection with said elements and said means for regulating the flow of air passing through said slots to render the latter effective when said elements are tilted above normal angles.

Patent No. 1,724,669. *Rigid Airship With Separate Gas Cells.* Hugo Eckener, Friedrichshafen-on-the-Bodensee, Germany, assignor to Luftschiffbau Zeppelin Gesellschaft mit beschränkter Haftung, Friedrichshafen-on-the-Bodensee, Germany. Three claims.

An airship comprising a cell filled with a non-inflammable carrying gas, another cell filled with inflammable gas, a collapsible cell within and laterally communicating with said former cell and means connected with said latter cell for allowing the escape of inflammable gas overboard.

Patent No. 1,724,135. *Aircraft.* Elbert G. Reid, Hampton, Va. Eight claims.

In combination, a body designed to cruise by its motion through a fluid, means having components perpendicular to the direction of motion, and fins secured to said body in planes substantially perpendicular to the mean cross stream dimensions of the said body, said fins having portions extending above and below said body in dimensions which are at any point substantially proportional to the difference between the pressure at that point and the pressure of the undisturbed fluid.

Patent No. 1,724,666. *Mooring Means for Airships.* Ralph H. Upson, Detroit, Michigan, assignor to Aircraft Development Corporation, Detroit, Michigan. Eight claims.

Airship mooring equipment comprising a ground post with attaching means substantially flush with the surrounding ground and means adapted to depress the post laterally, a turn-tide position about said post and two circular tracks about said post as a center, having they ground gear associated to resist load and lift and adapted to float with the ship and the line mooring.

Patent No. 1,725,914. *Device for Propelling Aircraft at High Altitudes by Direct Fluid Reaction.* Edwin Halliwell, Haverford, near Leeds, England. Eight claims.

Patent No. 1,721,665. *Pusher Airplane.* Carl G. Thompson, St. Louis, Mo. Seventeen claims.

Patent No. 1,724,765. *Airplane.* Vincent J. Burnell, New York, N. Y. One claim.

Patent No. 1,721,176. *Airplane.* Randolph P. Hall, Ithaca, N. Y., assignor of one-fourth to Theodore F. Hall, Wallingford, Conn., and one-fourth to Paul W. Adams, Ithaca, N. Y. Nineteen claims.

Patent No. 1,723,002. *Aircraft Engine Mount.* Harold Gorman, Garden City, N. Y., assignor to Fairchild-Canton Engine Corp., New York, N. Y. Eleven claims.

Patent No. 1,721,375. *Master Connecting Rod for Radial-Cylinder Engines.* Edward T. Jones, Ridgewood, N. J., assignor to Wright Aeronautical Corp. Five claims.

Patent No. 1,723,925. *Airplane-Propelling Means.* Sherman M. Cowell, New York, N. Y. One claim.

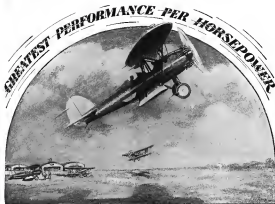
Patent No. 1,721,943. *Airplane Slack Absorber.* Frederick R. Weymouth, Hampton, N. Y., assignor to Fairchild Aircraft Mfg. Corp., New York, N. Y. Seven claims.

Thirteen Designs Receive Department A.T.C.

During the period of 30 days, August 21 to September 21, thirteen designs have received Aeronautics Type Certificates issued by the Aeronautics Branch, Department of Commerce. The addition of the newly licensed machines brings the total of licensed planes up to 222. The designs granted certificates during the 30 day period are products of General, Boeing, Stearns, Curtiss, Cessna, Robertson, Command-Aire, Kreider-Resener, New Standard, Looming and Fokker.

Complete specifications of new designs will be found in the supplement of the Aeronautical Engineering Section.

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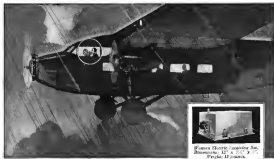
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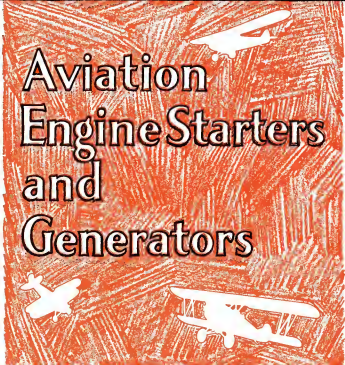
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